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Hygro-Thermal Vibrations and Buckling of laminated nanoplates via Nonlocal Strain Gradient Theory

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Abstract

Vibrations and buckling of thin laminiated composite nano plates in hygrothermal environment are investigated using second-order strain gradient theory. Hamilton's principle is used in order to carry out motion equations. To obtain analytical solution Navier displacement field has been considered for both cross- and angle-ply laminates. Numerical solutions are provided and discussed in terms of plate aspect ratio and non local ratio for a large number of laminates. Whenever possible a comparison with classical analytical solutions is reported for buckling loads and fundamental frequencies. This work shows a large variety of angle-ply cases which are not common in the published literature. Moreover, critical temperatures for cross- and angle-ply laminates are shown for buckling and free vibration analyses.

Keywords: Kirchhoff plate's theory, Non-local theory, Strain gradient theory, Hygrothermal load, Buckling, Free vibration, Composite nanoplates, Cross- and Angle-ply laminates

1. Introduction

In the last decades MEMS (Micro-Electro-Mechanical-System) and NEMS (Nano-Electro-Mechanical-System) have become topics of great interest because of their large number of applications in many industrial fields [1, 2, 3, 4]. These kind of structures, such as nanoplates, nanorods, nanobeams, can be

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used in medicine [5], electronics [6], aerospace [7] and even in civil construction [8]. To properly describe the behaviour of nanostructures it is necessary to use theories that take into account the nano size effect, like long range atomic interaction [9, 10]. Effects at the nano scale have been experimentally measured in [11, 12]. Non-local theories have been widely used for the study of nanostructures since Eringen developed his theory of non-local elasticity [13]. These theories consider the nano scale effects thanks to the introduction of one or more length scale parameters in addition to well know linear elastic Lamé parameters [14, 15, 16, 17]. The classification of nonlocal theories is generally presented as: strain gradient [18, 19, 20, 21], stress gradient [22], modified strain gradient [23, 24, 25], couple stress [26], modified couple stress [27, 28], integral type [29, 30] and micropolar [31, 32, 33]. In [34] strain and stress gradient non local theory is used to study dynamic and buckling problems of elastic nanobeams. Nanoplates subjected to hygrothermal loads were also investigated in the works [35, 36, 37, 38], using different non-local theories. In [39, 40] the influence of the non-local parameter on the critical load is studied and the solution for the problem of buckling of ccomposite nanoplates is provided. In [41] different non-local theories were employed to model the vibrational behavior of plates. Civalek et al. [42] presented numerical studies for dimensionless natural frequencies of different truss and frame models, investigating the influences of the nonlocal parameter. In [43], thermally induced dynamic behaviors of functionally graded flexoelectric nanobeams (FGFNs) are analyzed using semplified strain gradient nonlocal theory. The effect of thermal, hygrometric and piezoelectric stress on composite plates and shells has been investigated by [44, 45, 46, 47].

The focus of this paper is the study of buckling and free vibrations of laminated composite nano plates in hygrothermal environment. In particular, for the buckling analysis it will look for the temperature value and the combination of temperature and humidity that leads to the instability of the plate, while for the dynamic case it will investigate the influence that the thermal load has on the natural vibration frequencies. This paper is structured as described below. After the introduction section, the theoretical background for laminated thin plates in hygrothermal environment is developed introducing also the non-linear terms of von Karman that allow to perform the linear analysis of buckling. Using second order strain gradient theory non local effect are take into account. The analytical solution is obtained using Navier developments in double trigonometric series. Then, in order to validate the calculation code, implemented in MATLAB, various comparisons with the

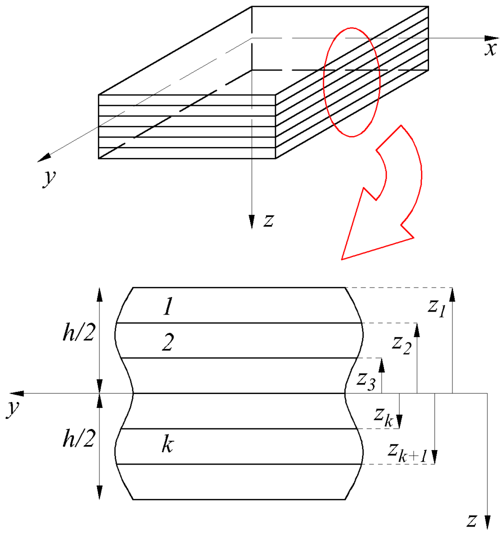

Figure 1: Laminate general layout.

literature are reported [48, 49, 50, 51]. After the comparisons, the results for buckling and free vibration obtained for different lamination schemes and different types of load are provided. Finally, a conclusion section is reported at the end of this paper.

2. Theoretical background

Consider a laminated thin nanoplate, modeled with the Kirchhoff plate assumptions modified to take into account the non linear terms of von Karman, subjected to hygrothermal stresses. The plate is composed of k orthotropic layers oriented at angles $\theta^{(1)}$, $\theta^{(2)}$, ..., $\theta^{(k)}$. The thickness of the k-th oriented layer, along the z axis, is defined as $h_k = z_{k+1} - z_k$. Introduced the reference system as in figure 1, we can define the displacement field of a generic point of the solid by means of the triad of displacement components U, V, W, which are functions of the coordinates (x, y, z).

$$\begin{aligned} U(x,y,z,t) &= u(x,y,t) - z\frac{\partial w}{\partial x} \\ V(x,y,z,t) &= v(x,y,t) - z\frac{\partial w}{\partial y} \\ W(x,y,z,t) &= w(x,y,t) \end{aligned} \tag{1}$$

where u,v and w are the displacements along the x, y and z axis of the point on the middle surface and $\partial w/\partial x$ and $\partial w/\partial y$ are the corresponding rotations. The plate strains are defined as:

$$\boldsymbol{\varepsilon} = \boldsymbol{\varepsilon}^{(0)} + z\boldsymbol{\varepsilon}^{(1)} \tag{2}$$

where

$$\boldsymbol{\varepsilon}^{(0)} = \begin{Bmatrix} \varepsilon_{xx}^{(0)} \\ \varepsilon_{yy}^{(0)} \\ \gamma_{xy}^{(0)} \end{Bmatrix} = \begin{Bmatrix} \frac{\partial u}{\partial x} + \frac{1}{2}\left(\frac{\partial w}{\partial x}\right)^2 \\ \frac{\partial v}{\partial y} + \frac{1}{2}\left(\frac{\partial w}{\partial y}\right)^2 \\ \frac{\partial u}{\partial y} + \frac{\partial v}{\partial x} + \frac{\partial w}{\partial x}\frac{\partial w}{\partial y} \end{Bmatrix}, \quad \boldsymbol{\varepsilon}^{(1)} = \begin{Bmatrix} \varepsilon_{xx}^{(1)} \\ \varepsilon_{yy}^{(1)} \\ \gamma_{xy}^{(1)} \end{Bmatrix} = \begin{Bmatrix} -\frac{\partial^2 w}{\partial x^2} \\ -\frac{\partial^2 w}{\partial y^2} \\ -2\frac{\partial^2 w}{\partial x \partial y} \end{Bmatrix} \tag{3}$$

In order to take into account non local effects, the second order strain gradient theory is introduced as follows

$$\begin{aligned} \begin{Bmatrix} \sigma_{xx} \\ \sigma_{yy} \\ \sigma_{xy} \end{Bmatrix}^{(k)} &= \left(1 - \ell^2 \nabla^2\right) \begin{bmatrix} \bar{Q}_{11} & \bar{Q}_{12} & \bar{Q}_{16} \\ \bar{Q}_{12} & \bar{Q}_{22} & \bar{Q}_{26} \\ \bar{Q}_{16} & \bar{Q}_{26} & \bar{Q}_{66} \end{bmatrix}^{(k)} \begin{Bmatrix} \varepsilon_{xx} \\ \varepsilon_{yy} \\ \gamma_{xy} \end{Bmatrix} \\ &- \begin{bmatrix} \bar{Q}_{11} & \bar{Q}_{12} & \bar{Q}_{16} \\ \bar{Q}_{12} & \bar{Q}_{22} & \bar{Q}_{26} \\ \bar{Q}_{16} & \bar{Q}_{26} & \bar{Q}_{66} \end{bmatrix}^{(k)} \begin{Bmatrix} \alpha_{xx}\Delta T + \beta_{xx}\Delta C \\ \alpha_{yy}\Delta T + \beta_{yy}\Delta C \\ 2\alpha_{xy}\Delta T + 2\beta_{xy}\Delta C \end{Bmatrix}^{(k)} \end{aligned} \tag{4}$$

where the subscript $^{(k)}$ indicates the k-th orthotropic lamina, ℓ is the nonlocal parameter and the operator $\nabla^2 = \partial^2/\partial x^2 + \partial^2/\partial y^2$.

A linear variation of hygrothermal loads along the thickness is assumed:

$$\begin{aligned} \Delta T &= T_0 + zT_1/h \\ \Delta C &= C_0 + zC_1/h \end{aligned} \tag{5}$$

It is underlined that the $\bar{Q}_{ij}^{(k)}$ represent the engineering constants oriented towards the reference system of the problem [48]. The hygrothermal properties of each ply have to be oriented also:

$$\begin{aligned} \boldsymbol{\alpha}^{(k)} &= \begin{Bmatrix} \alpha_{xx} \\ \alpha_{yy} \\ 2\alpha_{xy} \end{Bmatrix}^{(k)} = \begin{Bmatrix} \alpha_1^{(k)} \cos^2\theta^{(k)} + \alpha_2^{(k)} \sin^2\theta^{(k)} \\ \alpha_1^{(k)} \sin^2\theta^{(k)} + \alpha_2^{(k)} \cos^2\theta^{(k)} \\ 2\left(\alpha_1^{(k)} - \alpha_2^{(k)}\right) \sin\theta^{(k)} \cos\theta^{(k)} \end{Bmatrix} \\ \boldsymbol{\beta}^{(k)} &= \begin{Bmatrix} \beta_{xx} \\ \beta_{yy} \\ 2\beta_{xy} \end{Bmatrix}^{(k)} = \begin{Bmatrix} \beta_1^{(k)} \cos^2\theta^{(k)} + \beta_2^{(k)} \sin^2\theta^{(k)} \\ \beta_1^{(k)} \sin^2\theta^{(k)} + \beta_2^{(k)} \cos^2\theta^{(k)} \\ 2\left(\beta_1^{(k)} - \beta_2^{(k)}\right) \sin\theta^{(k)} \cos\theta^{(k)} \end{Bmatrix} \end{aligned} \tag{6}$$

By integrating the stresses along the thickness we obtain:

$$
\begin{aligned}
\mathbf{N} = \begin{Bmatrix} N_{xx} \\ N_{yy} \\ N_{xy} \end{Bmatrix} &= \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} \begin{Bmatrix} \sigma_{xx} \\ \sigma_{yy} \\ \sigma_{xy} \end{Bmatrix}^{(k)} dz \\
\mathbf{M} = \begin{Bmatrix} M_{xx} \\ M_{yy} \\ M_{xy} \end{Bmatrix} &= \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} \begin{Bmatrix} \sigma_{xx} \\ \sigma_{yy} \\ \sigma_{xy} \end{Bmatrix}^{(k)} z \; dz
\end{aligned}
\tag{7}
$$

Such definition of stress resultants allows to define **A**,**D** and **B** matrices, called *membrane stiffness matrix, bending stiffness matrix* and *bending-membrane coupling stiffness matrix* [48], and vectors $\mathbf{A}^{\alpha}$, $\mathbf{A}^{\beta}$,$\mathbf{B}^{\alpha}$, $\mathbf{B}^{\beta}$, $\mathbf{D}^{\alpha}$ and $\mathbf{D}^{\beta}$ containing the hygrothermal properties of the laminate

$$
\begin{aligned}
\mathbf{A}^{\alpha} &= \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\alpha}^{(k)} dz = \sum_{k=1}^{N_L} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\alpha}^{(k)} \left(z_{k+1} - z_k \right) \\
\mathbf{B}^{\alpha} &= \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\alpha}^{(k)} z \; dz = \frac{1}{2} \sum_{k=1}^{N_L} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\alpha}^{(k)} \left(z_{k+1}^2 - z_k^2 \right) \\
\mathbf{D}^{\alpha} &= \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\alpha}^{(k)} z^2 \; dz = \frac{1}{3} \sum_{k=1}^{N_L} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\alpha}^{(k)} \left(z_{k+1}^3 - z_k^3 \right)
\end{aligned}
\tag{8}
$$

$$
\begin{aligned}
\mathbf{A}^{\beta} &= \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\beta}^{(k)} dz = \sum_{k=1}^{N_L} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\beta}^{(k)} \left(z_{k+1} - z_k \right) \\
\mathbf{B}^{\beta} &= \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\beta}^{(k)} z \; dz = \frac{1}{2} \sum_{k=1}^{N_L} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\beta}^{(k)} \left(z_{k+1}^2 - z_k^2 \right) \\
\mathbf{D}^{\beta} &= \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\beta}^{(k)} z^2 dz = \frac{1}{3} \sum_{k=1}^{N_L} \bar{\mathbf{Q}}^{(k)} \boldsymbol{\beta}^{(k)} \left(z_{k+1}^3 - z_k^3 \right)
\end{aligned}
\tag{9}
$$

the stress characteristics take the following form:

$$\begin{Bmatrix} N_{xx} \\ N_{yy} \\ N_{xy} \end{Bmatrix} = \left(1-\ell^2\nabla^2\right)\left(\begin{bmatrix} A_{11} & A_{12} & A_{16} \\ A_{12} & A_{22} & A_{26} \\ A_{16} & A_{26} & A_{66} \end{bmatrix}\begin{Bmatrix} \varepsilon_{xx}^{(0)} \\ \varepsilon_{yy}^{(0)} \\ \gamma_{xy}^{(0)} \end{Bmatrix} + \begin{bmatrix} B_{11} & B_{12} & B_{16} \\ B_{12} & B_{22} & B_{26} \\ B_{16} & B_{26} & B_{66} \end{bmatrix}\begin{Bmatrix} \varepsilon_{xx}^{(1)} \\ \varepsilon_{yy}^{(1)} \\ \gamma_{xy}^{(1)} \end{Bmatrix}\right)$$
$$- \begin{Bmatrix} A_1^\alpha \\ A_2^\alpha \\ A_6^\alpha \end{Bmatrix} T_0 - \begin{Bmatrix} B_1^\alpha \\ B_2^\alpha \\ B_6^\alpha \end{Bmatrix} \frac{1}{h} T_1 - \begin{Bmatrix} A_1^\beta \\ A_2^\beta \\ A_6^\beta \end{Bmatrix} C_0 - \begin{Bmatrix} B_1^\beta \\ B_2^\beta \\ B_6^\beta \end{Bmatrix} \frac{1}{h} C_1 \tag{10}$$

$$\begin{Bmatrix} M_{xx} \\ M_{yy} \\ M_{xy} \end{Bmatrix} = \left(1-\ell^2\nabla^2\right)\left(\begin{bmatrix} B_{11} & B_{12} & B_{16} \\ B_{12} & B_{22} & B_{26} \\ B_{16} & B_{26} & B_{66} \end{bmatrix}\begin{Bmatrix} \varepsilon_{xx}^{(0)} \\ \varepsilon_{yy}^{(0)} \\ \gamma_{xy}^{(0)} \end{Bmatrix} + \begin{bmatrix} D_{11} & D_{12} & D_{16} \\ D_{12} & D_{22} & D_{26} \\ D_{16} & D_{26} & D_{66} \end{bmatrix}\begin{Bmatrix} \varepsilon_{xx}^{(1)} \\ \varepsilon_{yy}^{(1)} \\ \gamma_{xy}^{(1)} \end{Bmatrix}\right)$$
$$- \begin{Bmatrix} B_1^\alpha \\ B_2^\alpha \\ B_6^\alpha \end{Bmatrix} T_0 - \begin{Bmatrix} D_1^\alpha \\ D_2^\alpha \\ D_6^\alpha \end{Bmatrix} \frac{1}{h} T_1 - \begin{Bmatrix} B_1^\beta \\ B_2^\beta \\ B_6^\beta \end{Bmatrix} C_0 - \begin{Bmatrix} D_1^\beta \\ D_2^\beta \\ D_6^\beta \end{Bmatrix} \frac{1}{h} C_1 \tag{11}$$

In order to carry out the equations of motion the Hamilton's principle is employed

$$\int_0^T \left(\delta U + \delta V - \delta K\right)\, dt = 0 \tag{12}$$

where δU is the virtual strain energy, δV is the virtual work done by the applied forces and δK is the virtual kinetic energy. Developing the terms in equation (12) the Hamilton's principle takes the following form:

$$\int_0^T \left[\int_{\mathcal{A}} \left[\begin{Bmatrix} \delta u_{,x} \\ \delta u_{,y} \\ \delta v_{,x} \\ \delta v_{,y} \\ \delta w_{,xx} \\ \delta w_{,yy} \\ \delta w_{,xy} \end{Bmatrix}^\top \begin{bmatrix} \mathcal{T}_{11} & \mathcal{T}_{12} & \mathcal{T}_{13} \\ \mathcal{T}_{21} & \mathcal{T}_{22} & \mathcal{T}_{23} \\ \mathcal{T}_{31} & \mathcal{T}_{32} & \mathcal{T}_{33} \\ \mathcal{T}_{41} & \mathcal{T}_{42} & \mathcal{T}_{43} \\ \mathcal{T}_{51} & \mathcal{T}_{52} & \mathcal{T}_{53} \\ \mathcal{T}_{61} & \mathcal{T}_{62} & \mathcal{T}_{63} \\ \mathcal{T}_{71} & \mathcal{T}_{72} & \mathcal{T}_{73} \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \end{Bmatrix} - \begin{Bmatrix} \delta w_{,x} & \delta w_{,y} \end{Bmatrix} \begin{bmatrix} \hat{N}_{xx} & \hat{N}_{xy} \\ \hat{N}_{xy} & \hat{N}_{yy} \end{bmatrix} \begin{Bmatrix} w_{,x} \\ w_{,y} \end{Bmatrix} \right.\right.$$
$$\left.\left. \begin{Bmatrix} \delta \ddot{u} \\ \delta \ddot{v} \\ \delta \ddot{w} \\ \delta \ddot{w}_{,x} \\ \delta \ddot{w}_{,y} \end{Bmatrix}^\top \begin{bmatrix} I_0 & 0 & 0 & -I_1 & 0 \\ 0 & I_0 & 0 & 0 & -I_1 \\ 0 & 0 & I_0 & 0 & 0 \\ -I_1 & 0 & 0 & I_2 & 0 \\ 0 & -I_1 & 0 & 0 & I_2 \end{bmatrix} \begin{Bmatrix} \delta u \\ \delta v \\ \delta w \\ \delta w_{,x} \\ \delta w_{,y} \end{Bmatrix} \right] dxdy \right] dt$$
$$+ \text{ boundary integral terms} = 0 \tag{13}$$

where the variational form of the displacement field is identified by δ, while its corresponding derivatives in time by the dots, the terms $\mathcal{T}_{ij}$ are shown in [52], $\hat{N}_{xx}$, $\hat{N}_{yy}$ and $\hat{N}_{xy}$ (defined in eq. (10)) identify the axial and shear buckling terms, including hygrothermal terms, and I_0,I_1 and I_2 are the mass inertias which can be defined as it follows:

$$I_i = \rho \sum_{k=1}^{N_L} \int_{z_k}^{z_{k+1}} z^i \, dz \tag{14}$$

where $i = 0, 1, 2$.

3. Navier solution

The Navier solution is ontained for cross- and angle-ply laminates. This kind of solution allows to solve the case of simply supported plates [48].

For cross-ply laminates it is needed that $A_{16} = A_{26} = B_{16} = B_{26} = D_{16} =$

$D_{26} = 0$, and the displacement field it is assumed to be:

$$
\begin{aligned}
u(x,y) &= \sum_{m=1}^{\infty}\sum_{n=1}^{\infty} U_{mn} \cos\alpha x \sin\beta y \\
v(x,y) &= \sum_{m=1}^{\infty}\sum_{n=1}^{\infty} V_{mn} \sin\alpha x \cos\beta y \\
w(x,y) &= \sum_{m=1}^{\infty}\sum_{n=1}^{\infty} W_{mn} \sin\alpha x \sin\beta y
\end{aligned} \tag{15}
$$

For angle-ply laminates it is needed that $A_{16} = A_{26} = B_{11} = B_{12} = B_{22} = B_{66} = D_{16} = D_{26} = 0$,and the displacement field it is assumed to be:

$$
\begin{aligned}
u(x,y) &= \sum_{m=1}^{\infty}\sum_{n=1}^{\infty} U_{mn} \sin\alpha x \cos\beta y \\
v(x,y) &= \sum_{m=1}^{\infty}\sum_{n=1}^{\infty} V_{mn} \cos\alpha x \sin\beta y \\
w(x,y) &= \sum_{m=1}^{\infty}\sum_{n=1}^{\infty} W_{mn} \sin\alpha x \sin\beta y
\end{aligned} \tag{16}
$$

It is remarked the cross- and angle-ply laminated consider different kind of simply-supported boundary conditions [48].

As described in [48] shear in-plane mechanical load $\hat{N}_{xy} = 0$ should be neglected to solve the problem with Navier method. Since hygrothermal loads consider all the in-plane loads coupled, contrary to the mechanical in-plane loads, it is necessary that the lamination scheme for angle-ply plates is anti-symmetric so that it gives $\hat{N}_{xy} = 0$ (see eq. (10)). All cross-ply configurations have always $\hat{N}_{xy} = 0$.

3.1. Buckling

In this paragraph, the behavior of the plates subjected to thermal loads that lead to the instability will be analyzed. The solution system is:

$$
\begin{bmatrix} \hat{c}_{11} & \hat{c}_{12} & \hat{c}_{13} \\ \hat{c}_{12} & \hat{c}_{22} & \hat{c}_{23} \\ \hat{c}_{13} & \hat{c}_{23} & \hat{c}_{33} + \tilde{t}_{buck} \end{bmatrix} \begin{Bmatrix} U_{mn} \\ V_{mn} \\ W_{mn} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \\ 0 \end{Bmatrix} \tag{17}
$$

where the coefficients $\hat{c}_{ij}$ are those shown in [52] for the cross- and angle-ply plates, and the term $\tilde{t}_{buck}$ includes hygrothermal loads

$$\begin{aligned}\tilde{t}_{buck} = \bar{T}\Bigg(& \alpha^2 \left(A_1^\alpha + \kappa_{c_0} A_1^\beta + \frac{\kappa_t}{h} B_1^\alpha + \frac{\kappa_{c_1}}{h} B_1^\beta \right) \\ & + \beta^2 \left(A_2^\alpha + \kappa_{c_0} A_2^\beta + \frac{\kappa_t}{h} B_2^\alpha + \frac{\kappa_{c_1}}{h} B_2^\beta \right) \Bigg)\end{aligned} \tag{18}$$

where $\kappa_{c_0} = C_0/T_0$, $\kappa_t = T_1/T_0$ and $\kappa_{c_1} = C_1/T_0$.

From this last relation it can be deduced that the instability analysis cannot be performed with the load acting in one direction only because the type of load in question cannot be decoupled in two directions. As for the critical load, the critical temperature will be the lowest among the temperatures that lead to the instability of the plate.

$$T_{cr} = \min_{1 \le m,n \le \infty} \left\{ \bar{T}(m,n) \right\} \tag{19}$$

3.2. Free vibration

Replacing the Navier displacement field in the equations of motion and neglecting the rotary inertia, in the dynamic case, we obtain the following eigenvalue problem;

$$\left(\begin{bmatrix} \hat{c}_{11} & \hat{c}_{12} & \hat{c}_{13} \\ \hat{c}_{12} & \hat{c}_{22} & \hat{c}_{23} \\ \hat{c}_{13} & \hat{c}_{23} & \hat{c}_{33} + \tilde{t}_{buck} \end{bmatrix} - \omega^2 \begin{bmatrix} \hat{m}_{11} & 0 & 0 \\ 0 & \hat{m}_{22} & 0 \\ 0 & 0 & \hat{m}_{33} \end{bmatrix} \right) \begin{Bmatrix} U_{mn} \\ V_{mn} \\ W_{mn} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \\ 0 \end{Bmatrix} \tag{20}$$

where:

$$\begin{aligned} \hat{m}_{11} &= \hat{m}_{22} = I_0 \\ \hat{m}_{33} &= I_0 + I_2 \left(\alpha^2 + \beta^2 \right) \end{aligned} \tag{21}$$

For a nontrivial solution the determinant of the coefficient matrix should be zero, which yields to the characteristic polynomial. The real positive roots of this cubic equation give the square of the natural frequency associated with mode (m, n). The smallest of the frequencies is called the fundamental frequency. In the present case, the applied hygrothermal pre-stress influences the stiffness of the structure, thus, natural frequencies are obtained as a function of the pre-stress. There exist a value of the applied pre-stress that leads to null natural frequencies, such temperature values are defined as critical temperatures.

4. Results and discussion

In this section critical temperatures for the buckling and free vibration problems are discussed. Results are compared with the existing literature to validate the present model and novel applications are reported in order to demonstrate the influence of nanoscale parameter on the buckling and free vibration modes.

4.1. Buckling

The results of the first comparison are listed in table 1 with respect to the work [50], for a thin structure made of a single isotropic layer and a single orthotropic layer. For the isotropic configuration it has been considered: $E = 10^6$, $\nu = 0.3$, $\alpha_1/\alpha_0 = 1$, $\alpha_2/\alpha_0 = 1$, whereas for the orthotropic one: $E_1 = 15$, $E_2 = E_3 = 1$, $\nu_{12} = 0.3$, $\nu_{13} = 0.49$, $\nu_{23} = 0.3$, $G_{12} = 0.5$, $G_{13} = 0.3356$, $\alpha_1/\alpha_0 = 0.015$, $\alpha_2/\alpha_0 = 1$ where the normalization factor $\alpha_0 = 10^{-6}$ is taken into consideration. The results are presented in dimensionless form according to the formula $\alpha_0 T_{cr} \cdot 10^3$, where T_{cr} is the critical temperature that leads the buckling into buckling mode.

Lamina	a/h	Ref. [50]	Present
Isotropic	100	0.1265	0.1265
Orthotropic	100	0.7480	0.7486

Table 1: $\alpha_0 T_{cr} \cdot 10^3$ of a single square isotropic layer and a single square orthotropic layer compared with the literature $(m, n = 1)$.

Using the properties of the orthotropic layer of the first comparison laminates composed of multiple layers are analyzed. Results are shown in table 2. It is noted that the symmetric configuration (0/90/0) buckles with a non symmetric number of waves one along x and two along y ($m = 1$ and $n = 2$), on the contrary the antisymmetric scheme (0/90) has a symmetric buckling mode ($m = n = 1$). In both cases very good agreement is shown with the present implementation.

Another comparison has been performed with respect to the work by Shi et al [51] and comparison is listed in table 3. Mechanical properties of the plate considered are $a = 38.1$ cm, $b = 30.5$ cm, $h = 0.12$ cm, $E_1 = 155$ GPa, $E_2 = 8.07$ GPa, $G_{12} = 4.55$ GPa, $\nu_{12} = 0.22$, $\alpha_1 = -0.07 \cdot 10^{-6}$ °C^{-1}, $\alpha_2 = 30.1 \cdot 10^{-6}$ °C^{-1}. Very good agreement is shown considering that laboratory experiments have been carried out in [51].

Layout	a/h	Ref. [50]	Present
$(0/90)^a$	100	0.4860	0.4863
$(0/90/0)^b$	100	0.9960	0.9944

Table 2: $\alpha_0 T_{cr} \cdot 10^3$ of a square nano-plates compared with the literature $^a(m, n = 1)$, $^b(m = 1, n = 2)$.

Layout	a/h	Ref. [51]	Present
$(0/90/90/0)_s$	317.5	6.8 °C	6.575 °C

Table 3: T_{cr} of a cross-ply nanoplate compared with the literature $(m, n) = 1$.

4.1.1. In-plane thermal load

For the analysis of laminates with different values of the non-local parameter (tables 4, 5) the material properties are given as: $E_1/E_2 = var.$, $\nu_{12} = 0.25$, $\nu_{13} = \nu_{23} = 0$, $G_{12} = G_{13} = 0.5E_2$, $G_{23} = 0.2E_2$, $\alpha_1/\alpha_0 = 0.015$, $\alpha_2/\alpha_0 = 1$, $\beta_1 = 0$, $\beta_2 = 0.44$, where the stiffness ratio E_1/E_2 is variable and represents orthotropic material variation for a unitary in-plane transverse stiffness E_2.

Please note that the units are not reported because a consistent system has been implicitly considered. moreover, with the present selection the results are dimensionless for the thermal case only. Whereas, the results will be reported in factorized form for the hygrothermal applications. The plates considered are rectangular with a ratio $a/h = 100$ and the total height of the laminate is kept constant independently on the number of plies in each stack.

From tables 4, 5 it can be noticed that the critical temperature is higher for angle-ply laminates than for cross-ply ones with the same number of laminae. For antisymmetric cross- and angle-ply plates the instability always occurs for $(m, n) = (1, 1)$, whereas for symmetrical cross-ply plates the instability comes for different values of (m, n). In addition when the nonlocal parameter increases the critical temperature also increases.

Figure 2 displays different behaviours of cross- and angle-ply laminates by varying the geometric a/b and stiffness E_1/E_2 ratios for different values of the nonlocal parameter. It must be underlined that is considered $a = 1$ for the variation a/b, since such results are not reported in dimentionless form but they are factorized by $\alpha_0 T_{cr}$. The critical buckling temperatures increase almost exponentially by enlarging the plate width in the direction transverse

$(\ell/a)^2$	a/b	E_1/E_2 5	10	20	25	40
				(0/90/0)		
0.00	0.5	$0.5246^{(1,2)}$	$0.7522^{(1,3)}$	$1.0529^{(1,3)}$	$1.1855^{(1,3)}$	$1.5277^{(1,3)}$
	1.0	0.5246	0.7902	$1.1309^{(1,2)}$	$1.2413^{(1,2)}$	$1.5302^{(1,2)}$
	1.5	0.5803	0.7522	1.0529	1.1855	1.5277
0.05	0.5	0.9885	$1.5702^{(1,2)}$	$2.4710^{(1,2)}$	$2.8591^{(1,2)}$	$3.8346^{(1,2)}$
	1.0	1.0423	1.5702	2.4710	3.4050	3.8346
	1.5	1.5111	1.9587	2.7415	2.8591	3.9779
0.10	0.5	1.3656	$2.3501^{(1,2)}$	$3.6984^{(1,2)}$	$4.2792^{(1,2)}$	$5.7392^{(1,2)}$
	1.0	1.5601	2.3501	3.6984	4.2792	5.7392
	1.5	2.4418	3.1651	4.4301	4.9882	6.4280
				$(0/90)_2$		
0.00	0.5	0.3494	0.5357	0.8510	0.9864	1.3267
	1.0	0.4908	0.7549	1.0688	1.2250	1.6174
	1.5	0.8432	1.2477	1.9323	2.2265	2.9657
0.05	0.5	0.5651	0.8663	1.3759	1.5949	2.1450
	1.0	0.9752	1.4018	2.1238	2.4341	3.2137
	1.5	2.1956	3.2489	5.0314	5.7975	7.7221
0.10	0.5	0.7807	1.1968	1.9008	2.2033	2.9634
	1.0	1.4597	2.0981	3.1787	3.6432	4.8101
	1.5	3.5480	5.2501	8.1304	9.3684	12.4785

Table 4: $\alpha_0 T_{cr} \cdot 10^3$ of different cross-ply laminates for different values of the geometric a/b and stiffness E_1/E_2 and the non-local parameter $(\ell/a)^2$. The superscripts indicate the number of semi-waves for which the plate becomes unstable $^{(m,n)}$, where $(m,n) = (1,1)$ is not indicated.

$(\ell/a)^2$	a/b	E_1/E_2 5	10	20	25	40
				$(-45/45)$		
0.00	0.5	0.2855	0.3580	0.4656	0.5102	0.6206
	1.0	0.4951	0.6418	0.8583	0.9478	1.1693
	1.5	0.7790	0.9967	1.3189	1.4521	1.7822
0.05	0.5	0.4617	0.5789	0.7528	0.8249	1.0034
	1.0	0.9839	1.2753	1.7054	1.8832	2.3232
	1.5	2.0283	2.5952	3.4340	3.7811	4.6404
0.10	0.5	0.6378	0.7997	1.0401	1.1397	1.3862
	1.0	1.4726	1.9088	2.5525	2.8186	3.4773
	1.5	3.2776	4.1938	5.5549	6.1102	7.4987
				$(-45/45)_2$		
0.00	0.5	0.3827	0.6096	0.9932	1.1581	1.5723
	1.0	0.6809	1.1271	1.8820	2.2063	3.0209
	1.5	1.0607	1.7302	2.8627	3.3493	4.5716
0.05	0.5	0.6188	0.9856	1.6059	1.8725	2.5421
	1.0	1.3529	2.2397	3.7395	4.3839	6.0025
	1.5	2.7617	4.5052	7.4539	8.7209	11.9035
0.10	0.5	0.8549	1.3616	2.2186	2.5869	3.5120
	1.0	2.0249	3.3522	5.5969	6.5614	8.9840
	1.5	4.4629	7.2801	12.0450	14.0925	19.2354

Table 5: $\alpha_0 T_{cr} \cdot 10^3$ of different angle-ply laminates for different values of the geometric a/b and stiffness E_1/E_2 and the non-local parameter $(\ell/a)^2$. $(m,n) = (1,1)$.

$(\ell/a)^2$	κ_t	E_1/E_2 5	10	20	25	40
0.00	0	0.8432	1.2477	1.9323	2.2265	2.9657
	5	0.9613	1.3916	2.0782	2.3584	3.0260
	10	1.1177	1.5730	2.2479	2.5068	3.0889
0.05	0	2.1956	3.2489	5.0314	5.7975	7.7221
	5	2.5029	3.6235	5.4112	6.1408	7.8792
	10	2.9103	4.0958	5.8532	6.5272	8.0428
0.10	0	3.5840	5.2501	8.1304	9.3684	12.4785
	5	4.0446	5.8554	8.7443	9.9231	12.7324
	10	4.7029	6.6186	9.4584	10.5477	12.9968

Table 6: $\alpha_0 T_{0,cr} \cdot 10^3$ of a rectangular plate ($a/b = 1.5$) with lamination layout $(0/90)_2$ for different value of ratio $\kappa_t = T_{1,cr}/T_{0,cr}$ and non local parameter $(\ell/a)^2$.$(m,n) = (1,1)$.

to the fibers (zero fiber angle corresponds to the x axis which is related to plate width a).

4.1.2. In-plane and bending thermal loads

The effect of constant and linear thermal loads is investigated below. The aim is to show the effect of a linear temperature field to the buckling of the plate, this effect is considered with the coefficient $\kappa_t = T_{1,cr}/T_{0,cr}$. Table 6 shows the buckling when combined thermal load for cross-ply laminates is considered. The plate is of rectangular shape $a/b = 1.5$ with $a = 1$ and anti-symmetric cross-ply configuration. It is clear that the buckling temperature increases as the linear temperature increases, this induces the plate to show a stiffer behavior. Such increase is observed by increasing the stiffness ratio E_1/E_2 and the nonlocal parameter. Figure 3 shows the different behavior of nanoplates when they are subjected to a uniform and linear combination of temperature along the thickness for different geometric ratios a/b with $a = 1$. It should be remarked that the effect of combining constant and linear temperature distributions does not effect the critical temperature for square plates since all the curves coincide for $a/b = 1$. Overall small effects on the critical temperature are observed by including a linear temperature distribution.

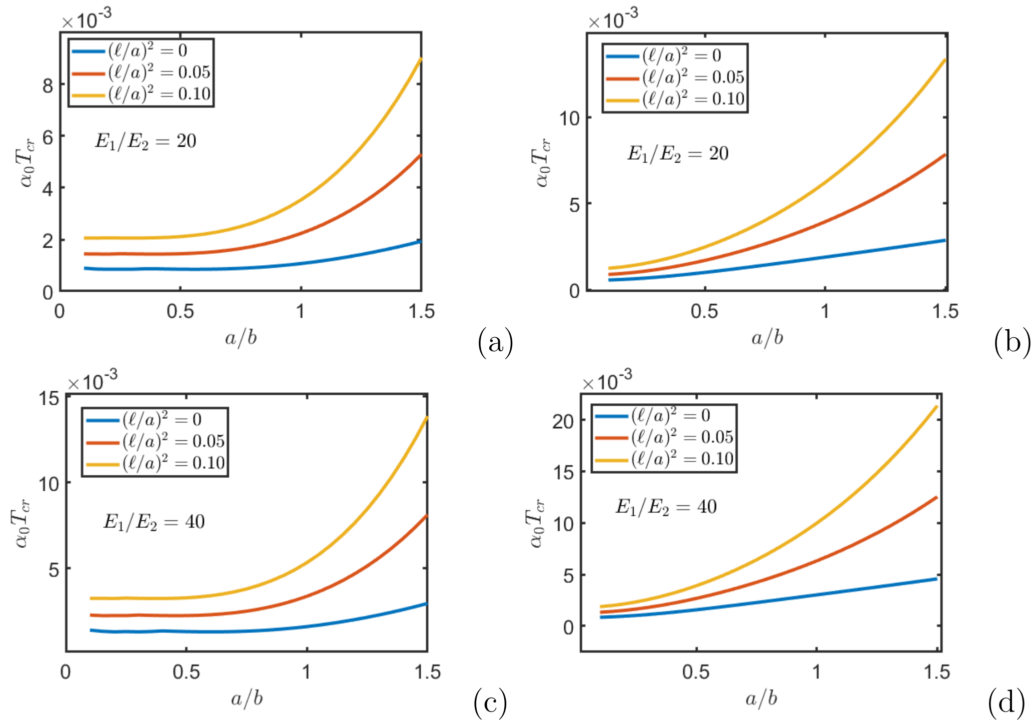

Figure 2: Critical temperature ($\alpha_0 T_{cr}$) of plates $(0/90)_2$ (a,c) and $(-45/45)_2$ (b,d) for different a/b and different value of non local parameter $(\ell/a)^2$.

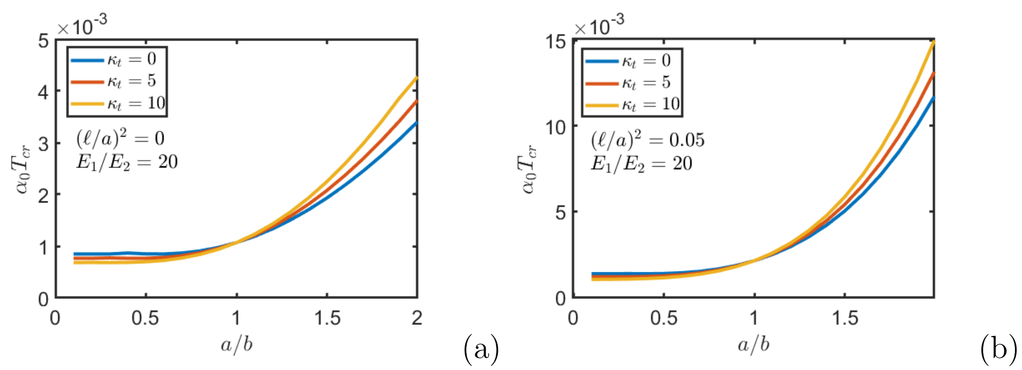

Figure 3: Critical temperature $\alpha_0 T_{cr}$ of plate with lamination layout $(0/90)_2$ for $(\ell/a)^2 = 0$ (a) and for $(\ell/a)^2 = 0.05$ (b) to vary of κ_t.

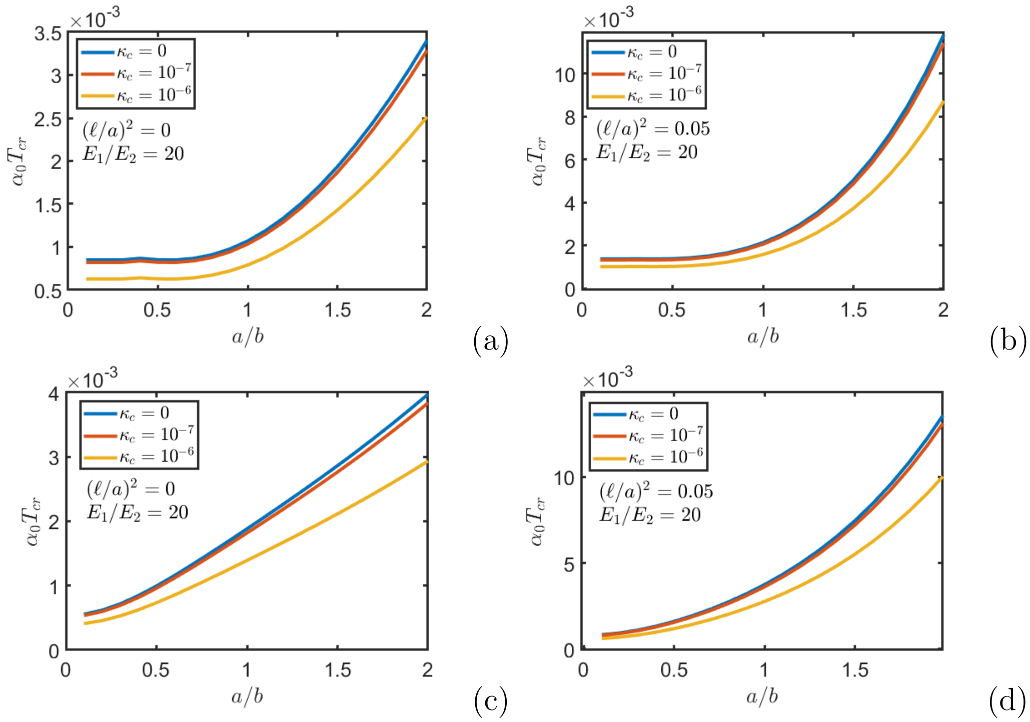

Figure 4: Critical temperature $\alpha_0 T_{cr}$ of plate by varying κ_{c_0} with lamination layout $(0/90)_2$ for $(\ell/a)^2 = 0$ (a) and for $(\ell/a)^2 = 0.05$ (b) with lamination layout $(-45/45)_2$ for $(\ell/a)^2 = 0$ (c) and for $(\ell/a)^2 = 0.05$ (d).

4.1.3. Hygrothermal loads

For hygrothermal a rectangular plate with ratio $a/b = 1.5$ with $a = 1$ is considered. Table 7 lists the combined buckling loads with $\kappa_{c_0} = C_{0,cr}/T_{0,cr}$ for both cross- and angle-ply laminates. Globally the critical temperature increases, as in the previous cases, by increasing the stiffness ratio E_1/E_2 and the nonlocal parameter. It is mentioned that by increasing the ratio $\kappa_{c_0} = C_{0,cr}/T_{0,cr}$, the critical temperature decreases because the critical load is weighted between the two values of $C_{0,cr}$ and $T_{0,cr}$. Figure 4 show the critical temperature for cross- and angle-play laminates, when they are subjected to hygrothermal load combination. It is noted that the angle-ply laminates have a smaller exponential increase with respect to the cross-ply ones by varying the geometric ratio a/b for $a = 1$.

$(\ell/a)^2$	κ_{c_0}	E_1/E_2				
		5	10	20	25	40
		$(0/90)_2$				
0.00	0	0.8432	1.2477	1.9323	2.2265	2.9657
	10^{-7}	0.8097	1.2007	1.8662	2.1538	2.8802
	10^{-6}	0.5964	0.8965	1.4271	1.6645	2.2871
0.05	0	2.1956	3.2489	5.0314	5.7975	7.7221
	10^{-7}	2.1083	3.1264	4.8594	5.6081	7.4996
	10^{-6}	1.5528	2.3343	3.7160	4.3340	5.9552
0.10	0	3.5840	5.2501	8.1304	9.3684	12.4785
	10^{-7}	3.4069	5.0521	7.8525	9.0624	12.1189
	10^{-6}	2.5093	3.7721	6.0048	7.0035	9.6233
		$(-45/45)_2$				
0.00	0	1.0607	1.7302	2.8627	3.3493	4.5716
	10^{-7}	1.0185	1.6650	2.1143	3.2399	4.4398
	10^{-6}	0.7502	1.2431	2.2479	2.5038	3.5255
0.05	0	2.7618	4.5052	7.4539	8.7209	11.9035
	10^{-7}	2.6520	4.3353	7.1990	8.4360	11.5605
	10^{-6}	1.9533	3.2369	5.5051	6.5194	9.1798
0.10	0	4.4629	7.2801	12.0451	14.0925	19.2354
	10^{-7}	4.2855	7.0056	11.6333	13.6322	18.6811
	10^{-6}	3.1564	5.2307	8.8960	10.5350	14.8341

Table 7: $\alpha_0 T_{0,cr} \cdot 10^3$ of a rectangular plate ($a/b = 1.5$) with lamination layout $(0/90)_2$ for different value of ratio $\kappa_{c_0} = C_{0,cr}/T_{0,cr}$ and non local parameter $(\ell/a)^2$. $(m,n) = (1,1)$.

4.2. Free Vibration

In this section results of free vibration, including thermal effects, are reported. The critical temperature will be analyzed, which corresponds to the temperature at which the natural frequency of free vibration becomes zero. The following material properties are used in the computations below: $E_1/E_2 = var.$, $\nu_{12} = 0.25$, $\nu_{13} = \nu_{23} = 0$, $G_{12} = G_{13} = 0.5E_2$, $G_{23} = 0.2E_2$, $\alpha_1/\alpha_0 = 1$, $\alpha_2/\alpha_0 = 3$. The plate is considered squared $a = b = 1$ for all numerical simulations. Natural frequencies are factorized as:

- cross-ply: $\bar{\omega} = \omega b^2/\pi^2\sqrt{\rho h/D_{22}}$
- angle-ply: $\bar{\omega} = \omega a^2/h\sqrt{\rho/E_2}$

Table 8 lists the results compared to [48, 52]. The results of the present work agree well with the ones presented in former literature but these results do not include any hygrothermal effect.

Figure 5 shows the influence of temperature in natural vibration frequencies of cross- and angle-ply laminates for different values of non local parameter and different lamination layouts. By reducing the temperature a detrimental effect is observed in the structural stiffness since the main natural frequency reduces garishly. There exists a temperature value for which the natural frequency is equal to zero, that temperature is called critical temperature for free vibrations. It can be noted that critical temperature for angle-ply laminates is higher than those of cross-ply laminates for plates with same number of laminae. In other words angle-ply laminates are able to vibrate at higher temperatures with respect to cross-ply before collapse. The same evidence in reported in tabular form in table 9 where critical temperatures for cross- and angle-ply laminates are provided.

5. Conclusions

In this paper, hygrothermal buckling and dynamic problems of simply supported composite nano plates were investigated. Non local second strain gradient theory is implemented for taking into account the effects of nano scale. Through Hamilton's principle motion equations for laminated composite thin plates are derived. The analytical solution using Navier solution method is obtained. Several plate layouts, materials and geometries are involved, comparisons for the classical case wherever it was possible are provided, then outcomes are extended to non local theory. Firstly outcomes for

Layout	$(\ell/a)^2$	Ref. [48]	Ref. [52]	Present
		$E_1/E_2 = 10$		
(0/90)	0	1.183	1.183	1.183
	0.05	—	1.668	1.668
	0.10	—	2.041	2.040
$(0/90)_4$	0	1.545	1.545	1.545
	0.05	—	2.178	2.177
	0.10	—	2.664	2.664
		$E_1/E_2 = 20$		
(0/90)	0	0.990	0.990	0.990
	0.05	—	1.395	1.395
	0.10	—	1.707	1.707
$(0/90)_4$	0	1.469	1.469	1.469
	0.05	—	2.071	2.071
	0.10	—	2.534	2.534
		$E_1/E_2 = 25$		
$(-45/45)$	0	12.357	12.358	12.357
	0.05	—	17.419	17.419
	0.10	—	21.311	21.310
$(-45/45)_4$	0	20.154	20.154	20.154
	0.05	—	28.409	28.409
	0.10	—	34.756	34.756
		$E_1/E_2 = 40$		
$(-45/45)$	0	14.636	14.636	14.636
	0.05	—	20.631	20.630
	0.10	—	25.241	25.239
$(-45/45)_3$	0	24.825	24.825	24.825
	0.05	—	34.994	34.994
	0.10	—	42.812	42.811

Table 8: Fundamental frequencies $\bar{\omega}$ for cross- and angle-ply laminates.

$(\ell/a)^2$	(0/90)	$(0/90)_4$	$(-45/45)$	$(-45/45)_4$
0	35.2506	77.6826	55.5655	138.4088
0.05	70.0422	154.3526	110.4064	275.0129
0.10	104.8334	231.0231	165.2473	411.6170

Table 9: Critical temperature $T_{0,cr}$ for cross- and angle-ply laminates with $E_1/E_2 = 20$.

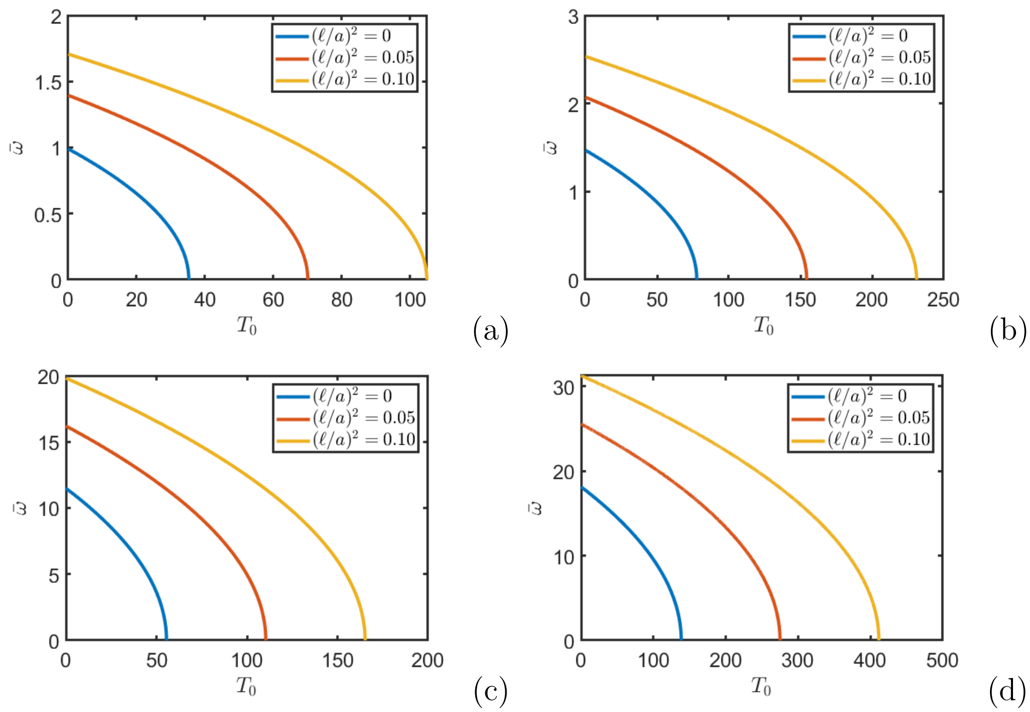

Figure 5: Natural frequency ($\bar{\omega}$) versus temperature (T_0) with $E_1/E_2 = 20$ for (a) (0/90) , (b) $(0/90)_4$, (c) $(-45/45)$ and (d) $(-45/45)_4$.

thermal and combined hygrothermal buckling of cross- and angle-ply laminates are provided, it can be seen that for the same number of laminae, angle-ply laminates are preferable, moreover for the same thickness is better to have more laminae. Finally outcomes for free vibration are reported. At first the classic problem is investigated and compared, then thermal terms are included and the critical temperatures for various lamination layouts and values of non local parameter is obtained. Also in this case, angle-ply laminates show a better behavior than cross-ply ones.

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