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Published Version:
Cornacchia, F., Fabbrocino, F., Fantuzzi, N., Luciano, R., Penna, R. (2021). Analytical solution of cross- and angle-ply nano plates with strain gradient theory for linear vibrations and buckling. MECHANICS OF ADVANCED MATERIALS AND STRUCTURES, 28(12), 1201-1215 [10.1080/15376494.2019.1655613].

Availability:
This version is available at: https://hdl.handle.net/11585/698729 since: 2024-09-19

Published:
DOI: http://doi.org/10.1080/15376494.2019.1655613

Analytical solution of cross- and angle-ply nano plates with strain gradient theory for linear vibrations and buckling

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Abstract

Vibrations and buckling of Kirchhoff nano plates are investigated using second-order strain gradient theory. The Navier displacement field has been considered for two different sets of boundary conditions and stacking sequences. Different geometries and material properties for isotropic, orthotropic cross- and angle-ply laminates are considered, and numerical simulations are discussed in terms of plate aspect ratio and non local ratio. A comparison with the classical analytical solution is provided whenever possible for buckling loads and fundamental frequencies.

Keywords: Stability analysis, Dynamic analysis, Orthotropic laminate, Nano-structures, Nonlocal elastic theory, Analytical modelling

1. Introduction

In the current literature MEMS (Micro-Electro-Mechanical-System) and NEMS (Nano-Electro-Mechanical-System) are topics of relevant interest be-

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Preprint submitted to Mechanics of Advanced Materials and Structures August 23, 2019

cause of their various uses [1, 2, 3]. Indeed, these types of materials can be employed in many areas of application, i.e. engineering, medicine and electronics [4, 2, 5, 6, 7], in the form of generators, transistors, sensors, actuators, resonators, detectors etc.

This work wants to focus the attention on NEMS, which are usually modeled by simulating small scale effects on nano rods, nano beams, nano tubes and nano plates. In fact, the mechanical behavior of nano structural components is size-dependent [8, 9, 10, 11, 12], highly influenced by the material structure and by the interactions at the atomic scale among particles at distant location, as commented in [13, 14, 15, 16, 17, 18, 19, 20, 21], effects that have much lower impact in macro structures. Thus, in order to take into account the size effects, the classical continuum mechanics theories are not suitable, which implies the application of modified versions [22, 23, 24, 25], that are based on the individuation of an internal length scale. A wide range of non classical theories have been developed in order to capture the non locality effects, among which Eringen [26, 26] was one of the pioneer and his nonlocal elasticity theory has been extensively applied in the study of nano structures by scientists [27, 28, 29]. Hence, an important milestone in the practice of higher order theories of linear elasticity is to determine the correct non local relation [30, 31]. A broad list of higher order theories of linear elasticity can be found in literature, among which,strain gradient, modified strain gradient, stress gradient, modified couple stress and micropolar theories can be identified [32**?** , 33, 25, 34, 35, 36, 37, 38, 39], and the choice depends on the research to carry out and on the ability of the scientists. Here, the effort will be focused on the development of studies of buckling and vibrations of nano plates, which is a relevant subject for the scientific community, as it can be found in [40, 41, 42, 43, 44, 45, 46, 47, 48, 49]. An easy theory, which is applied in the present study, is the second order-strain gradient theory that establish a connection between stress and strain of the structure in the constitutive equations through a single non local parameter, as previously done by Papargyri-Beskos [50]. The method followed in the present paper follows the one presented in [51] for static analysis of laminates, where the gap between the theories in terms of deflection and stresses is shown. In fact, the Kirchhoff governing equations in weak form are carried out by considering the size effects, while the Navier displacement field is applied in order to develop the analytical solution in terms of stability and dynamic analysis. Comparison with Reddy [52], Papargyri-Beskos [50] and Babu Patel [21] are provided if possible for the classical continuum mechanics theory, before extending the application to orthotropic laminated materials (cross- and angle-ply laminates) employing the second order-strain gradient theory.

2. Theoretical model

2.1. Kirchhoff theory

Different combinations of geometrical and material configurations of orthotropic thin rectangular nanoplates are implemented by making use of the classical laminated plate theory (CLPT). In order to conduct stability and dynamic analysis for such structures, at nano scale level, a modification of the theory, based on the bending plate hypothesis of Kirchhoff is needed. The laminates have dimension a and b along x- and y-axis, respectively, while the thickness of the generic oriented k-th lamina $h_k = z_{k+1} - z_k$, as it is displayed in Fig. 1 For the case of geometric non linearity, the displacements in the three directions can be written from the Kirchhoff assumptions and restrictions as it follows:

$$\begin{aligned} u(x,y,z,t) &= u_0(x,y,t) - z w_{0,x} \\ v(x,y,z,t) &= v_0(x,y,t) - z w_{0,y} \\ w(x,y,z,t) &= w_0(x,y,t) \end{aligned} \tag{1}$$

where, u_0, v_0, w_0 are the displacements along x-, y- and z-axis of the points on the mid-surface, and $w_{0,x}$ and $w_{0,y}$ are the homologous rotations.

The plate strain is expressed in the von Karman form:

$$\varepsilon = \left\{\varepsilon^{(m)}\right\} + z\left\{\varepsilon^{(f)}\right\} = \begin{Bmatrix} \varepsilon_{xx}^{(m)} \\ \varepsilon_{yy}^{(m)} \\ \gamma_{xy}^{(m)} \end{Bmatrix} + z \begin{Bmatrix} \varepsilon_{xx}^{(f)} \\ \varepsilon_{yy}^{(f)} \\ \gamma_{xy}^{(f)} \end{Bmatrix} \tag{2}$$

where, $^{(m)}$ indicates the membrane strain, while $^{(f)}$ the flexural strain. ε_{xx} and ε_{yy} are the normal strains along x and y directions respectively, instead γ_{xy} represents the in-plane shear strain. Consequently, the membrane and flexural strains can be written as function of the displacements:

$$\begin{Bmatrix} \varepsilon_{xx}^{(m)} \\ \varepsilon_{yy}^{(m)} \\ \gamma_{xy}^{(m)} \end{Bmatrix} = \begin{bmatrix} u_{0,x} + \frac{1}{2} w_{0,x}^{2} \\ v_{0,y} + \frac{1}{2} w_{0,y}^{2} \\ u_{0,y} + v_{0,x} + w_{0,x}\, w_{0,y} \end{bmatrix}, \quad \begin{Bmatrix} \varepsilon_{xx}^{(f)} \\ \varepsilon_{yy}^{(f)} \\ \gamma_{xy}^{(f)} \end{Bmatrix} = \begin{bmatrix} -w_{0,xx} \\ -w_{0,yy} \\ -2w_{0,xy} \end{bmatrix} \tag{3}$$

In order to take into account the effects of non locality due to the dimensions of the nano plates, the second-order strain gradient theory must

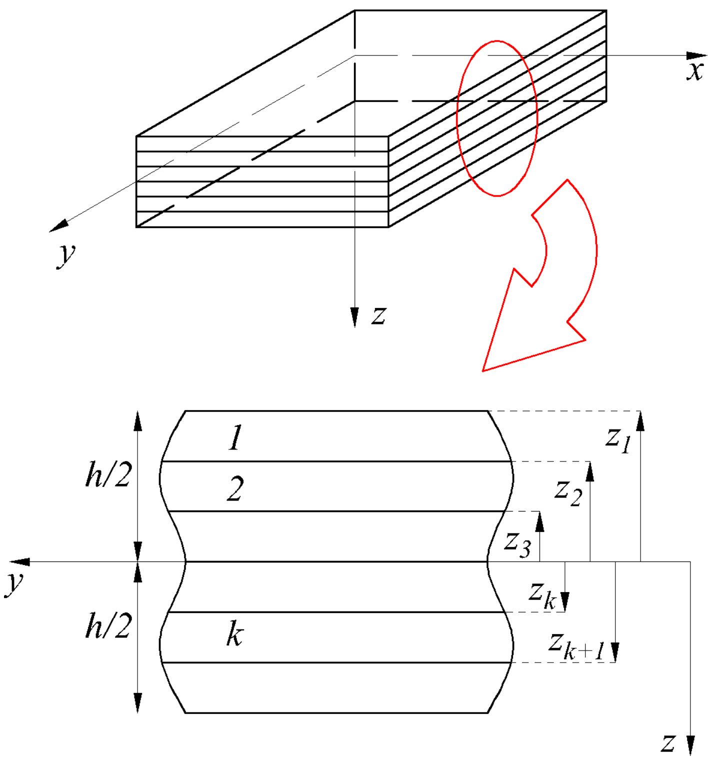


Figure 1: Laminate general layout

be involved in the computation. For the k-th orthotropic lamina in terms of laminate coordinates, the constitutive equations can be written as:

$$\begin{Bmatrix} \sigma_{xx} \\ \sigma_{yy} \\ \tau_{xy} \end{Bmatrix}^{(k)} = (1-\ell^2\nabla^2) \begin{bmatrix} \bar{Q}_{11} & \bar{Q}_{12} & \bar{Q}_{16} \\ \bar{Q}_{12} & \bar{Q}_{22} & \bar{Q}_{26} \\ \bar{Q}_{16} & \bar{Q}_{26} & \bar{Q}_{66} \end{bmatrix}^{(k)} \begin{Bmatrix} \varepsilon_{xx} \\ \varepsilon_{yy} \\ \gamma_{xy} \end{Bmatrix}^{(k)} \tag{4}$$

where, $\nabla^2 = \partial^2/\partial y^2 + \partial^2/\partial x^2$, and $\bar{Q}_{ij}$ are function of sheets orientations and are derived from the engineering constants in accordance with the formulations below:

$$\begin{aligned} Q_{11} &= \frac{E_1}{1-\nu_{12}\nu_{21}}, \quad Q_{22} = \frac{E_2}{1-\nu_{12}\nu_{21}} \\ Q_{12} &= \frac{E_1\nu_{21}}{1-\nu_{12}\nu_{21}} = \frac{E_2\nu_{12}}{1-\nu_{12}\nu_{21}}, \quad Q_{66} = G_{12} \end{aligned} \tag{5}$$

where, E_1, E_2 are the Young's moduli, ν_{12} and ν_{21} are the Poisson's ratii and G_{12} is the shear modulus.

The dynamic version of the principle of the virtual works (Hamilton's Principle) is employed in order to carry out the equations of motion. It is important to point out that the transverse shear stress, needed for the equilibrium of the plate, has been involved in the boundary conditions and equilibrium of forces.

$$\int_0^T (\delta U + \delta V - \delta K) = 0 \tag{6}$$

with, δU is the virtual strain energy, δV is the virtual work done by the applied forces, and δK is the virtual kinetic energy

Developing the terms in Eq. (6), the Hamilton's Principle can be conveniently written in extended matrix form as:

$$\int_0^T \int_{\Omega_0} \Bigg[\begin{Bmatrix} \delta u_{0,x} \\ \delta u_{0,y} \\ \delta v_{0,x} \\ \delta v_{0,y} \\ \delta w_{0,xx} \\ \delta w_{0,yy} \\ \delta w_{0,xy} \end{Bmatrix}^T \begin{bmatrix} \mathcal{T}_{11} & \mathcal{T}_{12} & \mathcal{T}_{13} \\ \mathcal{T}_{21} & \mathcal{T}_{22} & \mathcal{T}_{23} \\ \mathcal{T}_{31} & \mathcal{T}_{32} & \mathcal{T}_{33} \\ \mathcal{T}_{41} & \mathcal{T}_{42} & \mathcal{T}_{43} \\ \mathcal{T}_{51} & \mathcal{T}_{52} & \mathcal{T}_{53} \\ \mathcal{T}_{61} & \mathcal{T}_{62} & \mathcal{T}_{63} \\ \mathcal{T}_{71} & \mathcal{T}_{72} & \mathcal{T}_{73} \end{bmatrix} \begin{Bmatrix} u_0 \\ v_0 \\ w_0 \end{Bmatrix}$$

$$- \begin{Bmatrix} \delta w_{0,x} & \delta w_{0,y} \end{Bmatrix} \begin{bmatrix} \hat{N}_{xx} & \hat{N}_{xy} \\ \hat{N}_{xy} & \hat{N}_{yy} \end{bmatrix} \begin{Bmatrix} w_{0,x} \\ w_{0,y} \end{Bmatrix} \tag{7}$$

$$+ \begin{Bmatrix} \delta \ddot{u}_0 \\ \delta \ddot{v}_0 \\ \delta \ddot{w}_0 \\ \delta \ddot{w}_{0,x} \\ \delta \ddot{w}_{0,y} \end{Bmatrix}^T \begin{bmatrix} I_0 & 0 & 0 & -I_1 & 0 \\ 0 & I_0 & 0 & 0 & -I_1 \\ 0 & 0 & I_0 & 0 & 0 \\ -I_1 & 0 & 0 & I_2 & 0 \\ 0 & -I_1 & 0 & 0 & I_2 \end{bmatrix} \begin{Bmatrix} u_0 \\ v_0 \\ w_0 \\ w_{0,x} \\ w_{0,y} \end{Bmatrix} \Bigg] dxdy \Bigg] dt$$

$$+ \text{ boundary integral terms} = 0$$

where the variational form of the displacement field is dentified by δ, while its corresponding derivatives in time by the dots, the terms $\mathcal{T}$ are shown in the appendix, $\hat{N}_{xx}, \hat{N}_{yy}, \hat{N}_{xy}$ identify the axial and shear buckling terms and I_0, I_1, I_2 are the mass inertias which can be defined as it follows:

$$I_i = \rho \sum_{k=1}^{N} \int_{z_k}^{z_{k+1}} z^i \, dz \tag{8}$$

where, $i = 0, 1, 2$. The following resultants of forces and moments are obtained by integrating the stresses for each layer through the z-axis:

$$\begin{Bmatrix} N_{xx} \\ N_{yy} \\ N_{xy} \end{Bmatrix} = (1-\ell^2\nabla^2)\left(\begin{bmatrix} A_{11} & A_{12} & A_{16} \\ A_{12} & A_{22} & A_{26} \\ A_{16} & A_{26} & A_{66} \end{bmatrix} \begin{Bmatrix} \varepsilon_{xx}^{(m)} \\ \varepsilon_{yy}^{(m)} \\ \gamma_{xy}^{(m)} \end{Bmatrix} + \begin{bmatrix} B_{11} & B_{12} & B_{16} \\ B_{12} & B_{22} & B_{26} \\ B_{16} & B_{26} & B_{66} \end{bmatrix} \begin{Bmatrix} \varepsilon_{xx}^{(f)} \\ \varepsilon_{yy}^{(f)} \\ \gamma_{xy}^{(f)} \end{Bmatrix} \right) \tag{9}$$

$$\begin{Bmatrix} M_{xx} \\ M_{yy} \\ M_{xy} \end{Bmatrix} = (1-\ell^2\nabla^2)\left(\begin{bmatrix} B_{11} & B_{12} & B_{16} \\ B_{12} & B_{22} & B_{26} \\ B_{16} & B_{26} & B_{66} \end{bmatrix} \begin{Bmatrix} \varepsilon_{xx}^{(m)} \\ \varepsilon_{yy}^{(m)} \\ \gamma_{xy}^{(m)} \end{Bmatrix} + \begin{bmatrix} D_{11} & D_{12} & D_{16} \\ D_{12} & D_{22} & D_{26} \\ D_{16} & D_{26} & D_{66} \end{bmatrix} \begin{Bmatrix} \varepsilon_{xx}^{(f)} \\ \varepsilon_{yy}^{(f)} \\ \gamma_{xy}^{(f)} \end{Bmatrix} \right) \tag{10}$$

where, the stiffnesses are computed as it follows:

$$\begin{aligned} A_{ij} &= \sum_{k=1}^{N} \bar{Q}_{ij}^{(k)}(z_{k+1} - z_k) \\ B_{ij} &= \frac{1}{2}\sum_{k=1}^{N} \bar{Q}_{ij}^{(k)}(z_{k+1}^2 - z_k^2) \\ D_{ij} &= \frac{1}{3}\sum_{k=1}^{N} \bar{Q}_{ij}^{(k)}(z_{k+1}^3 - z_k^3) \end{aligned} \tag{11}$$

The linear equations of motion of the classical laminated plate theory in terms of displacement, accounting for non local effects are obtained by setting the non linear terms equal to zero and by carrying out integration by parts in (7):

$$
\begin{aligned}
&A_{11}u_{0},_{xx} + 2A_{16}u_{0},_{xy} + A_{66}u_{0},_{yy} + A_{16}v_{0},_{xx} + (A_{12} + A_{66})v_{0},_{xy} + A_{26} \\
&v_{0},_{yy} - \big[B_{11}w_{0},_{xxx} + 3B_{16}w_{0},_{xxy} + (B_{12} + 2B_{66})w_{0},_{xyy} + B_{26}w_{0},_{yyy}\big] - \ell^{2}\big[A_{11} \\
&\big(u_{0},_{xxxx} + u_{0},_{xxyy}\big) + 2A_{16}\big(u_{0},_{xxxy} + u_{0},_{xyyy}\big) + A_{66}\big(u_{0},_{xxyy} + u_{0},_{yyyy}\big) + A_{16} \\
&\big(v_{0},_{xxxx} + v_{0},_{xxyy}\big) + (A_{12} + A_{66})\big(v_{0},_{xxxy} + v_{0},_{xyyy}\big) + A_{26}\big(v_{0},_{xxyy} + v_{0},_{yyyy}\big) - \\
&- \big[B_{11}\big(w_{0},_{xxxxx} + w_{0},_{xxxyy}\big) + 3B_{16}\big(w_{0},_{xxxxy} + w_{0},_{xxyyy}\big) + (B_{12} + 2B_{66}) \\
&\big(w_{0},_{xxxyy} + w_{0},_{xyyyy}\big) + B_{26}\big(w_{0},_{xxyyy} + w_{0},_{yyyyy}\big)\big]\big] = I_{0}\ddot{u}_{0} - I_{1}\ddot{w}_{0},_{x}
\end{aligned}
\tag{12}
$$

$$
\begin{aligned}
&A_{16}u_{0},_{xx} + (A_{12} + A_{66})u_{0},_{xy} + A_{26}u_{0},_{yy} + A_{66}v_{0},_{xx} + 2A_{26}v_{0},_{xy} + A_{22} \\
&v_{0},_{yy} - \big[B_{16}w_{0},_{xxx} + (B_{12} + 2B_{66})w_{0},_{xxy} + 3B_{26}w_{0},_{xyy} + B_{22}w_{0},_{yyy}\big] - \ell^{2}\big[A_{16} \\
&\big(u_{0},_{xxxx} + u_{0},_{xxyy}\big) + (A_{12} + A_{66})\big(u_{0},_{xxxy} + u_{0},_{xyyy}\big) + A_{26}\big(u_{0},_{xxyy} + u_{0},_{yyyy}\big) + \\
&+ A_{66}\big(v_{0},_{xxxx} + v_{0},_{xxyy}\big) + 2A_{26}\big(v_{0},_{xxxy} + v_{0},_{xyyy}\big) + A_{22}\big(v_{0},_{xxyy} + v_{0},_{yyyy}\big) - \\
&- \big[B_{16}\big(w_{0},_{xxxxx} + w_{0},_{xxxyy}\big) + (B_{12} + 2B_{66})\big(w_{0},_{xxxxy} + w_{0},_{xxyyy}\big) + 3B_{26} \\
&\big(w_{0},_{xxxyy} + w_{0},_{xyyyy}\big) + B_{22}\big(w_{0},_{xxyyy} + w_{0},_{yyyyy}\big)\big]\big] = I_{0}\ddot{v}_{0} - I_{1}\ddot{w}_{0},_{y}
\end{aligned}
\tag{13}
$$

$$
\begin{aligned}
& B_{11}u_{0,xxx} + 3B_{16}u_{0,xxy} + (B_{12} + 2B_{66})u_{0,xyy} + B_{26}u_{0,yyy} + B_{16}v_{0,xxx} + (B_{12}+ \\
& + 2B_{66})v_{0,xxy} + 3B_{26}v_{0,xyy} - B_{22}v_{0,yyy} - \big[D_{11}w_{0,xxxx} + 4D_{16}w_{0,xxxy} + 2(D_{12}+ \\
& + 2D_{66})w_{0,xxyy} + 4D_{26}w_{0,xyyy} + D_{22}w_{0,yyyy}\big] - \ell^2\big[B_{11}\big(u_{0,xxxxx} + u_{0,xxxyy}\big)+ \\
& + 3B_{16}\big(u_{0,xxxxy} + u_{0,xxyyy}\big) + (B_{12} + 2B_{66})\big(u_{0,xxxyy} + u_{0,yyyy}\big) + B_{26} \\
& \big(u_{0,xxyyy} + u_{0,yyyyy}\big) + B_{16}\big(v_{0,xxxxx} + v_{0,xxxyy}\big) + (B_{12} + 2B_{66})\big(v_{0,xxxxy} + \\
& + v_{0,xxyyy}\big) + 3B_{26}\big(v_{0,xxxyy} + v_{0,xyyyy}\big) + B_{22}\big(v_{0,xxyyy} + v_{0,yyyyy}\big) - \big[D_{11} \\
& \big(w_{0,xxxxxx} + w_{0,xxxxyy}\big) + 4D_{16}\big(w_{0,xxxxxy} + w_{0,xxxyyy}\big) + 2(D_{12} + 2D_{66}) \\
& \big(w_{0,xxxxyy} + w_{0,xxyyyy}\big) + 4D_{26}\big(w_{0,xxxyyy} + w_{0,xyyyyy}\big) + D_{22}\big(w_{0,xxyyyy} + \\
& + w_{0,yyyyyy}\big)\big]\big] = I_1\big(\ddot{u}_{0,x} + \ddot{v}_{0,y}\big) + I_0\ddot{w}_0 - I_2\big(\ddot{w}_{0,xx} + \ddot{w}_{0,yy}\big) - \big(\hat{N}_{xx}w_{0,xx} + \\
& + 2\hat{N}_{xy}w_{0,xy} + \hat{N}_{yy}w_{0,yy}\big)
\end{aligned}
\tag{14}
$$

2.2. *Navier solution*

In this section, the Navier procedure for simply supported laminates is applied to orthotropic cross ply and angle ply laminates. By replacing the Navier displacement field, which will be made explicit in the corresponding subsections, in the system below (omitting the von Karman non linear terms) the analytical solutions are obtained:

$$
\begin{bmatrix} \hat{c}_{11} & \hat{c}_{12} & \hat{c}_{13} \\ \hat{c}_{12} & \hat{c}_{22} & \hat{c}_{23} \\ \hat{c}_{13} & \hat{c}_{23} & \hat{c}_{33} + \hat{s}_{33} \end{bmatrix}
\begin{Bmatrix} U_{mn} \\ V_{mn} \\ W_{mn} \end{Bmatrix}
+
\begin{bmatrix} \hat{m}_{11} & 0 & \hat{m}_{13} \\ 0 & \hat{m}_{22} & \hat{m}_{23} \\ \hat{m}_{13} & \hat{m}_{23} & \hat{m}_{33} \end{bmatrix}
\begin{Bmatrix} \ddot{U}_{mn} \\ \ddot{V}_{mn} \\ \ddot{W}_{mn} \end{Bmatrix}
=
\begin{Bmatrix} 0 \\ 0 \\ 0 \end{Bmatrix}
\tag{15}
$$

where, the terms in the matrices will be made explicit for cross- and angle-ply laminates.

The analytical solutions for the stability and dynamic analysis respectively, are carried out and shown below:

$$\bar{N} = \frac{1}{\alpha^2 + k\beta^2}\left(\hat{c}_{33} + \hat{c}_{13}\frac{a_1}{a_0} + \hat{c}_{23}\frac{a_2}{a_0}\right) \tag{16}$$

$$\bar{\omega}^2 = \frac{1}{\hat{m}_{33}}\left(\hat{c}_{33} + \hat{c}_{13}\frac{a_1}{a_0} + \hat{c}_{23}\frac{a_2}{a_0}\right) \tag{17}$$

where

$$\begin{aligned} a_{mn} &= \hat{c}_{33} + \hat{c}_{13}\frac{a_1}{a_0} + \hat{c}_{23}\frac{a_2}{a_0} \\ a_0 &= \hat{c}_{11}\hat{c}_{22} - \hat{c}_{12}\hat{c}_{12} \\ a_1 &= \hat{c}_{12}\hat{c}_{23} - \hat{c}_{13}\hat{c}_{22} \\ a_2 &= \hat{c}_{13}\hat{c}_{12} - \hat{c}_{11}\hat{c}_{23} \end{aligned} \tag{18}$$

2.2.1. Antisymmetric Cross-Ply Laminates

The Navier displacement field is assumed to be:

$$\begin{aligned} u_0(x,y) &= \sum_{n=1}^{\infty}\sum_{m=1}^{\infty} U_{mn}\cos\alpha x\sin\beta y \\ v_0(x,y) &= \sum_{n=1}^{\infty}\sum_{m=1}^{\infty} V_{mn}\sin\alpha x\cos\beta y \\ w_0(x,y) &= \sum_{n=1}^{\infty}\sum_{m=1}^{\infty} W_{mn}\sin\alpha x\sin\beta y \end{aligned} \tag{19}$$

where, $\alpha = m\pi/a$ and $\beta = n\pi/b$

in order to satisfy the displacement boundary conditions (SS-1), as it follows:

$$
\begin{aligned}
u_0(x,0,t)=0, u_0(x,b,t)=0, v_0(0,y,t)=0, v_0(a,y,t)=0 \\
w_0(x,0,t)=0, w_0(x,b,t)=0, w_0(0,y,t)=0, w_0(a,y,t)=0 \\
\left.\frac{\partial w_0}{\partial x}\right|_{(x,0,t)}=0, \left.\frac{\partial w_0}{\partial x}\right|_{(x,b,t)}=0, \left.\frac{\partial w_0}{\partial y}\right|_{(0,y,t)}=0, \left.\frac{\partial w_0}{\partial y}\right|_{(a,y,t)}=0,
\end{aligned} \tag{20}
$$

The coefficients to be used in Eq. (15), for the cross-ply laminate case are shown below:

$$
\begin{aligned}
\hat{c}_{11} =& -(\alpha^2 A_{11}+\beta^2 A_{66})-\ell^2[\alpha^4 A_{11}+\alpha^2\beta^2(A_{11}+A_{66})+\beta^4 A_{66}] \\
\hat{c}_{12} =& -\alpha\beta(A_{12}+A_{66})-\ell^2[\alpha^3\beta(A_{12}+A_{66})+\alpha\beta^3(A_{12}+A_{66})] \\
\hat{c}_{13} =& [\alpha^3 B_{11}+\alpha\beta^2(B_{12}+2B_{66})]+\ell^2[\alpha^5 B_{11}+\alpha^3\beta^2(B_{11}+B_{12}+2B_{66})+ \\
& +\alpha\beta^4(B_{12}+2B_{66})] \\
\hat{c}_{22} =& -(\alpha^2 A_{66}+\beta^2 A_{22})-\ell^2[\alpha^4 A_{66}+\alpha^2\beta^2(A_{22}+A_{66})+\beta^4 A_{22}] \\
\hat{c}_{23} =& [\beta^3 B_{22}+\alpha^2\beta(B_{12}+2B_{66})]+\ell^2[\beta^5 B_{22}+\alpha^2\beta^3(B_{22}+B_{12}+2B_{66})+ \\
& +\alpha^4\beta(B_{12}+2B_{66})] \\
\hat{c}_{33} =& -(\alpha^4 D_{11}+\beta^4 D_{22}+2\alpha^2\beta^2(D_{12}+2D_{66}))-\ell^2[\alpha^6 D_{11}+\beta^6 D_{22}+\alpha^4\beta^2 \\
& (D_{11}+2D_{12}+4D_{66})+\alpha^2\beta^4(D_{22}+2D_{12}+4D_{66})]
\end{aligned} \tag{21}
$$

$$
\begin{aligned}
\hat{m}_{11} &= \hat{m}_{22} = I_0 \\
\hat{m}_{13} &= -I_1\alpha \\
\hat{m}_{23} &= -I_1\beta \\
\hat{m}_{33} &= I_0 + I_2(\alpha^2+\beta^2) \\
\hat{s}_{33} &= (\alpha^2\hat{N}_{xx}+\beta^2\hat{N}_{yy})
\end{aligned} \tag{22}
$$

It is important to point out that, the solution for cross ply laminated with SS-1 boundary conditions is valid only if:

$$A_{16} = A_{26} = B_{16} = B_{26} = D_{16} = D_{26} = 0 \tag{23}$$

2.2.2. *Antisymmetric Angle-Ply Laminates*

The Navier displacement field for this case, is assumed to be:

$$\begin{aligned} u_0(x,y) &= \sum_{n=1}^{\infty}\sum_{m=1}^{\infty} U_{mn} \sin\alpha x \cos\beta y \\ v_0(x,y) &= \sum_{n=1}^{\infty}\sum_{m=1}^{\infty} V_{mn} \cos\alpha x \sin\beta y \\ w_0(x,y) &= \sum_{n=1}^{\infty}\sum_{m=1}^{\infty} W_{mn} \sin\alpha x \sin\beta y \end{aligned} \tag{24}$$

which satisfies the SS-2 boundary conditions:

$$\begin{aligned} u_0(0,y,t) = 0, u_0(a,y,t) = 0, v_0(x,0,t) = 0, v_0(x,b,t) = 0 \\ w_0(x,0,t) = 0, w_0(x,b,t) = 0, w_0(0,y,t) = 0, w_0(a,y,t) = 0 \\ \left.\frac{\partial w_0}{\partial x}\right|_{(x,0,t)} = 0, \left.\frac{\partial w_0}{\partial x}\right|_{(x,b,t)} = 0, \left.\frac{\partial w_0}{\partial y}\right|_{(0,y,t)} = 0, \left.\frac{\partial w_0}{\partial y}\right|_{(a,y,t)} = 0, \end{aligned} \tag{25}$$

where α and β are already defined in the previous subsection.

In this case, the coefficient to be employed in Eq. (15), are the following:

$$
\begin{aligned}
\hat{c}_{11} &= A_{11}\alpha^2 + A_{66}\beta^2 + \ell^2[A_{11}(\alpha^4 + \alpha^2\beta^2) + A_{66}(\beta^4 + \alpha^2\beta^2)] \\
\hat{c}_{12} &= (A_{12} + A_{66})\alpha\beta + \ell^2(A_{12} + A_{66})(\alpha\beta^3 + \alpha^3\beta) \\
\hat{c}_{13} &= -(3B_{16}\alpha^2\beta + B_{26}\beta^3) - \ell^2[3B_{16}(\alpha^4\beta + \alpha^2\beta^3) + B_{26}(\alpha^2\beta^3 + \beta^5)] \\
\hat{c}_{22} &= A_{66}\alpha^2 + A_{22}\beta^2 + \ell^2[A_{66}(\alpha^4 + \alpha^2\beta^2) + A_{22}(\beta^4 + \alpha^2\beta^2)] \\
\hat{c}_{23} &= -(B_{16}\alpha^3 + 3B_{26}\alpha\beta^2) - \ell^2[B_{16}(\alpha^5 + \alpha^3\beta^2) + 3B_{26}(\alpha\beta^4 + \alpha^3\beta^2)] \\
\hat{c}_{33} &= D_{11}\alpha^4 + 2(D_{12} + 2D_{66})\alpha^2\beta^2 + D_{22}\beta^4 + \ell^2[D_{11}(\alpha^6 + \alpha^4\beta^2) + 2(D_{12}+ \\
&\quad + 2D_{66})(\alpha^4\beta^2 + \alpha^2\beta^4) + D_{22}(\alpha^2\beta^4 + \beta^6)]
\end{aligned}
\tag{26}
$$

$$
\begin{aligned}
\hat{m}_{11} &= \hat{m}_{22} = I_0 \\
\hat{m}_{33} &= I_0 + I_2(\alpha^2 + \beta^2) \\
\hat{m}_{23} &= \hat{m}_{13} = 0 \\
\hat{s}_{33} &= (\alpha^2\hat{N}_{xx} + \beta^2\hat{N}_{yy})
\end{aligned}
\tag{27}
$$

Finally, the SS-2 boundary conditions exsist only if the stiffness:

$$
A_{16} = A_{26} = B_{11} = B_{12} = B_{22} = B_{66} = D_{16} = D_{26} = 0 \tag{28}
$$

3. Results - Stability analysis

3.1. Isotropic

Firstly, the outcomes for an isotropic single lamina were carried out in order to make the comparison with Papargyri et al. [50] for the case of buckling, assuming gradient elastic material behavior. The lamina is assumed to be simply supported, with the same dimensions along the x and y directions

($a = b$), while the properties of the isotropic material are: $E_1 = E_2 = E = 1$ (E_2 is always considered equal to one in the computations and E_1 will vary for cross- and angle-plies), $\nu = 0.25$ and $G = 0.5E/(1 + \nu)$. The solution in terms of buckling load, for uniaxial compression in x direction, which accounts for non locality effects, is dimensionless with respect to the classical solution ($\ell = 0$). Thus, in the graph below the dimensionless buckling load $\bar{N}$ is plotted as a function of the normalized gradient coefficient $(\ell/a)^2$, where the dots represent the solution of the Eq. (16), while the solid line is the computation of the reference equation from Ref. [50], obtained for $n = m = 1$ which correspond to the minimum value for square plates:

$$\bar{N} = \left[1 + 2\pi^2 \left(\frac{\ell}{a}\right)^2\right] \tag{29}$$

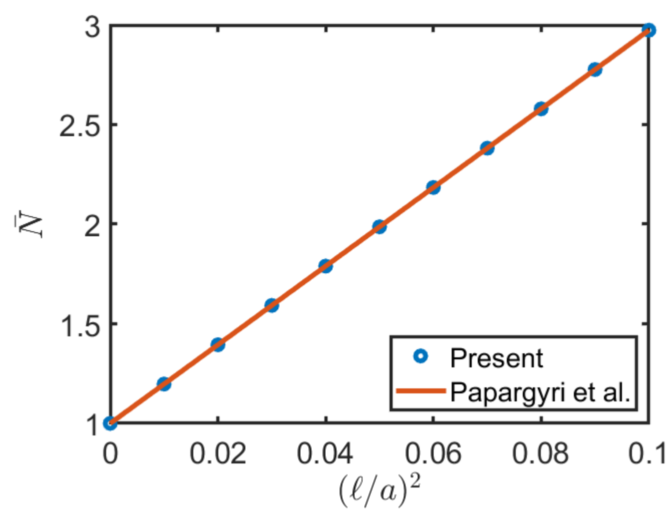
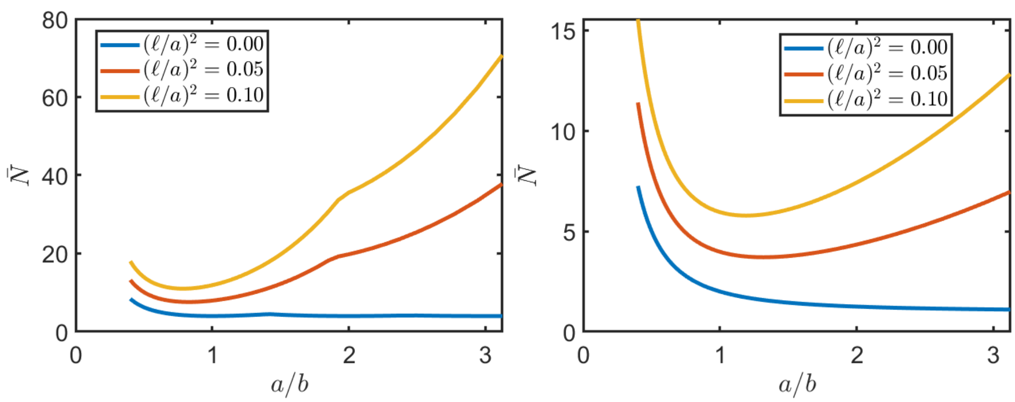


Figure 2: Buckling - comparison with Ref. [50].

The Figure 2 shows how in good agreement are the formulations. The rising trend displays that the critical buckling load grows with non local ratio $(\ell/a)^2$. Note that the minimum buckling load does not always occur for $m = n = 1$ for rectangular and laminated plate configurations as it will be discussed in the following. For this reason the minimum buckling load has been observed to occur within $m, n = 1, 2, 3$ in the present computations.

Once Eq. (16) has been verified, it is employed in order to understand the behavior to changing aspect ratios a/b. The material properties are the same as the previous case, beside the classical theory $(\ell/a)^2 = 0.00$, two more values of non local ratios are analized $(\ell/a)^2 = 0.05$ and $(\ell/a)^2 = 0.10$, while the compression is considered for uniaxial and biaxial cases, $k = 0$ and $k = 1$, respectively. It is important to point out that, using the Navier displacement field only the uniaxial and biaxial cases can be studied, while not the tangential buckling because the equations cannot work in this case.



a) b)

Figure 3: Nondimensionalized buckling load versus plate aspect ratio for isotropic lamina - a)Uniaxial compression, b)Biaxial compression

From the graph of uniaxial compression, it is possible to see how for the classical theory, after an initial decrising of dimensionless buckling load with a/b, the solution has stabilized behavior and quite smooth trend. On the other hand, if non local effects are involved in the computation, rising paths are shown since values lower than $a/b = 1$ after the initial decreasing. Moreover, discontinuities in both trends are displayed for a/b slight lower than two. From the second graph of Fig. 3, smooth paths are shown for the three cases and all of them have declining trend in the first phase. Then, the classical theory presents almost constant $\bar{N}$ since a/b around unity, while when the lamina is treated with second order theory, it answers with rising $\bar{N}$ to changing a/b.

In both cases, for the whole range of lamina dimensions taken into account, higher are the values of $(\ell/a)^2$ higher are the critical load magnitudes, moreover increasing gap between classical and non local theory to rising a/b are displayed.

3.2. Antisymmetric cross-ply

Secondly, othotropic cross-ply plates are studied. For the classical theory the comparison with Reddy [52] is provided whenever possible, then the application is extended to the second order theory, presenting outcomes for $(\ell/a)^2$ equal to 0.05 and 0.10. The ratio E_1/E_2 assumes different magnitudes, which will be given step by step, while $\nu_{12} = 0.25$, $G_{12} = G_{13} = 0.5E_2$ and $G_{23} = 0.2E_2$ are the same during the computation. In the first two sections of the tables 1 and 2, Reddy and present outcomes are reported for the classical theory, then the application is applied to $(\ell/a)^2$ equal to 0.05 and 0.10. The aspect ratios a/b treated are: 0.5, 1.0 and 1.5, while E_1/E_2 ratio assumes magnitudes equal to 5, 10, 20, 25 and 40, for $0/90/0/90 = (0/90)_2$ laminate layout. Buckling loads have been reported in dimensionless form as

	a/b	E_1/E_2 = 5	10	20	25	40
Reddy [52]	0.5	4.705	4.157	3.828	3.757	3.647
	1	2.643	2.189	1.923	1.866	1.778
	1.5	2.955	2.487	2.211	2.152	2.061
$(\ell/a)^2 = 0.00$	0.5	4.705	4.157	3.828	3.757	3.647
	1	2.643	2.189	1.923	1.866	1.778
	1.5	2.955	2.487	2.211	2.152	2.061
$(\ell/a)^2 = 0.05$	0.5	7.667	6.778	6.234	6.115	5.927
	1	5.422	4.546	3.994	3.868	3.661
	1.5	8.952	7.769	6.968	6.772	6.441
$(\ell/a)^2 = 0.10$	0.5	10.600	9.374	8.623	4.232	8.199
	1	8.131	6.830	6.0138	2.281	5.516
	1.5	14.500	12.617	11.340	3.340	10.486

Table 1: Uniaxial buckling loads ($k = 0$) for $(0/90)_2$ laminate configuration

it follows: $\bar{N} = N_{cr}[b^2/(\pi^2 D_{22})]$, considering as maximum order of expansion $m, n = 1, 2, 3$ because the critical buckling load was sought. Table 1 is referred to uniform uniaxial compression ($k = 0$), instead table 2 to biaxial one ($k = 1$).

In both Tab. 1 and 2 it is possible to see how results match accurately in the classic application, whereas as it was expected an increasing of the magnitude of the buckling loads is shown for the second order gradient theory. Moreover, it is complicated to make a comparison in terms of variable E_1/E_2 and a/b parameter, due to the fluctuating trends within the same theory,

		E_1/E_2				
	a/b	5	10	20	25	40
Reddy [52]	0.5	3.764	3.325	3.062	3.005	2.917
	1	1.322	1.095	0.962	0.933	0.889
	1.5	1.009	0.860	0.773	0.754	0.725
$(\ell/a)^2 = 0.00$	0.5	3.764	3.325	3.062	3.005	2.917
	1	1.322	1.095	0.962	0.933	0.889
	1.5	1.009	0.860	0.773	0.754	0.725
$(\ell/a)^2 = 0.05$	0.5	6.134	5.423	4.987	4.892	4.742
	1	2.711	2.273	1.997	1.934	1.830
	1.5	2.754	2.390	2.144	2.084	1.982
$(\ell/a)^2 = 0.10$	0.5	8.480	7.499	6.899	6.767	6.559
	1	4.065	3.415	3.007	2.913	2.758
	1.5	4.462	3.882	3.489	3.393	3.226

Table 2: Biaxial buckling loads ($k = 1$) for $(0/90)_2$ laminate configuration

especially for $k = 0$. Thus, in order to draw conclusions it is needed to represent outcomes in graphical form, wherein material properties chosen for the analysis are: $E_1/E_2 = 25$ and $E_1/E_2 = 40$. The laminate configurations studied are: (0/90), $(0/90)_2$ and $(0/90)_4$ for the uniform uniaxial compression ($k = 0$), while for the biaxial case (0/90), $(0/90)_2$ and $(0/90)_3$ are taken into account. The dimensionless expression used, is again: $\bar{N} = N_{cr}[b^2/(\pi^2 D_{22})]$, with a maximum expansion order of $m, n = 1, 2, 3$.

In Fig. 4 and 5, it is possible to see how the classical theory displays the lower critical loads for every laminate configuration, material and uniform compression type. It shows also discontinuities for the uniaxial compression, instead smooth trends for biaxial one, because the buckling load is not given by $m = n = 1$ for rectangular plates as discussed in classical references [52]. Moreover, for both classical and second gradient order theories, an initial reduction of the buckling load is shown, in the first case it is followed by a quite constant path, while for the second case it is visible the growing magnitude with increasing value of aspect ratio, where slope expands with non local ratios. Laminae made by the same sequence of layers, but accounting for different materials are studied and it comes out that if $E_1/E_2 = 25$ is considered as property of the material, an higher magnitude of buckling load is displaced compared to $E_1/E_2 = 40$ case. From the comparison among different layouts for both uniaxial and biaxial compression, it comes out that, to parity of materials and plate thickness, the lower critical load belongs to (0/90) configuration, while its value grows as more layers are added to the plate.

	E_1/E_2	10	25	40
$(45/-45)$	Reddy [52]	9.066	15.476	21.709
	$(\ell/a)^2 = 0.00$	9.066	15.476	21.709
	$(\ell/a)^2 = 0.05$	18.015	30.750	43.135
	$(\ell/a)^2 = 0.10$	26.963	46.024	64.561
$(45/-45)_4$	Reddy [52]	17.637	41.163	64.683
	$(\ell/a)^2 = 0.00$	17.637	41.163	64.683
	$(\ell/a)^2 = 0.05$	35.043	81.789	128.522
	$(\ell/a)^2 = 0.10$	52.450	122.415	192.362

Table 3: Uniaxial buckling loads for $(45/-45)$ and $(45/-45)_4$ laminate configurations

3.3. Antisymmetric angle-ply

Finally, in this section orthotropic angle-ply laminates are studied. As in the previous case in tables 3 and 4, the first two sections are referred to the comparison with Reddy of the classical theory [52], then it is extended to the second-order strain gradient theory. All the parameters employed can be picked from the previous paragraph, except the laminates taken into account which are: $(45/-45)$ and $(45/-45)_4$, while the E_1/E_2 ratios are specified in the tables. Both uniform uniaxial ($k = 0$) and biaxial ($k = 1$) compression of the square plate, along x, and x and y are carried out, using the following dimensionless expression: $\bar{N} = N_{cr}[b^2/(h^3E_2)]$.

From both Tab 3 and 4, the comparison for the classical theory leads to good confidence in the method also for orthotropic antisymmetric angle-ply laminates. Moreover, as in the earlier case it is possible to see an increasing in magnitude of the dimensionless buckling load to rising $(\ell/a)^2$. Consequently,

	E_1/E_2	10	25	40
$(45/-45)$	Reddy [52]	4.533	7.738	10.854
	$(\ell/a)^2 = 0.00$	4.533	7.738	10.854
	$(\ell/a)^2 = 0.05$	9.007	15.375	21.567
	$(\ell/a)^2 = 0.10$	13.481	23.012	32.280
$(45/-45)_4$	Reddy [52]	8.818	20.581	32.341
	$(\ell/a)^2 = 0.00$	8.818	20.581	32.341
	$(\ell/a)^2 = 0.05$	17.522	40.895	64.261
	$(\ell/a)^2 = 0.10$	26.225	61.208	96.181

Table 4: Biaxial buckling loads for $(45/-45)$ and $(45/-45)_4$ laminate configurations

as it follows, a deeper study in order to catch the trend of $(-45/45)_i$, (with $i = 1,2,3,4$) laminate configurations is carried out, enlarging the range of a/b up to five and considering E_1/E_2 equal to 25 and 40, in both $k = 0$ and $k = 1$ conditions.

As for the cross-ply laminates, the figures 6 and 7 show higher values of dimensionless buckling load for values of non local ratio equal to 0.10. Also, when $k = 0$ the classical theory displays flat trends, differently from the second-order strain gradient theory which presents rough tendency, instead if $k = 1$ they are always smooth. The behavior in case of $(\ell/a)^2 = 0$ presents an original decreasing followed by a stable trend, viceversa if $(\ell/a)^2$ is non zero the consecutive part grows up to very high values. In addition, comparing the different behavior of the plates it is possible to assert that to parity of material, the six-layered plate shows much higher critical load for every a/b, and comparing the two-, four- and six-layered laminate in $k = 0$ and $k = 1$

it is possible to see that for uniaxial case the structures buckles for much higher values.

4. Results - Dynamic analysis

4.1. Isotropic

Accordingly to what previously done for the stability analysis, also for the free vibration analysis the first step is to compare the present solution to the Papargyri et al. [50], which is expressed by Eq.(30) for isotropic materials. The material properties, of the square plate ($a = b$), are the following: $E_1/E_2 = 1$, $\nu = 0.25$ and $G = 0.5E/(1+\nu)$. The dimensionless frequency $\bar{\omega}$

$$\bar{\omega} = \sqrt{1 + 2\pi^2\left(\frac{\ell}{a}\right)^2} \tag{30}$$

has been plotted for changing dimensionless $(\ell/a)^2$, for $n = m = 1$. In Fig. 8 it is possible to see how outcomes match accurately, where the dots represent the solution of the Eq. (17), and the solid line is referred to the computation of Eq. (30), showing a rising parabolic behavior for the range of $(\ell/a)^2$ within 0 and 0.1. Thus, the study has been extended in order to understand the behavior for different plate geometries. In fact, outcomes are plotted in Fig. 9 considering an isotropic lamina, for non local ratios equal to 0, 0.05 and 0.10.

It is possible to see an increasing of the dimensionless frequency magnitude with $(\ell/a)^2$, for the whole path, showing higher gaps among theories as a/b rises. Moreover, the initial decreasing is followed by a stable trend for the classical theory ($(\ell/a)^2 = 0$) and by a rising one for the second-order strain gradient theory.

4.2. Antisymmetric cross-ply

Then, analysis continues facing to antisymmetric cross-ply laminates. Whenever it has been possible, comparisons with Reddy [52] are carried out for $(\ell/a)^2 = 0$, then results are extended to second-order strain gradient theory. In table 5, dimensionless frequencies of square antisymmetric cross-ply laminates (layouts: (0/90), $(0/90)_2$ and $(0/90)_4$) are carried out imposing $m, n = 1, 2, 3$. The comparison with Reddy [52] is provided in its first two sections for the classical theory, then the theory has been developed also for $(\ell/a)^2$ equal to 0.05 and 0.10. The material properties are given: E_1/E_2 equal to 10 and 20, $\nu_{12} = 0.25$, $G_{12} = G_{13} = 0.5E_2$ and $G_{23} = 0.2E_2$. The frequency is dimensionless with respect to the following formula: $\bar{\omega} = \omega b^2/\pi^2\sqrt{\rho h/D_{22}}$.

In Tab. 5 is it possible to see how results are in good agreement for what concerns the classical theory. Moreover, $\bar{\omega}$ increases with the number of layers in the laminate accounting for the same total thickness, for every mode and value of non local ratio. Thus, graphic results are drawn, for (0/90), $(0/90)_2$ and $(0/90)_4$ configurations, employing $m, n = 1, 2, 3$. Fundamental frequency is carried out with respect to the aspect ratio a/b, for magnitude of non local ratio $(\ell/a)^2$ equal to 0.00, 0.05 and 0.10. Materials selected are given by E_1/E_2 equal to 25 and 40, $\nu_{12} = 0.25$, $G_{12} = G_{13} = 0.5E_2$.

In Fig. 10, for the classical theory case, it is possible to see a reducing magnitude of dimensionless fundamental frequency which stabilizes for values of a/b between 1 and 2, for every geometrical configuration and material property. This is similar in the initial stage for second-order strain gradient

E_1/E_2			10			20		
	m	n	(0/90)	$(0/90)_2$	$(0/90)_4$	(0/90)	$(0/90)_2$	$(0/90)_4$
Reddy [52]	1	1	1.183	1.479	1.545	0.990	1.386	1.469
	1	2	3.174	4.077	4.274	2.719	3.913	4.158
	1	3	6.666	8.698	9.136	5.789	8.456	8.998
	2	1	3.174	4.077	4.274	2.719	3.913	4.158
	2	2	4.733	5.918	6.179	3.959	5.547	5.877
	2	3	7.927	10.034	10.494	6.702	9.507	10.088
	3	1	6.666	8.698	9.136	5.789	8.456	8.998
	3	2	7.927	10.034	10.494	6.193	9.507	10.088
	3	3	10.650	13.317	13.904	8.908	12.481	13.224
$(\ell/a)^2 = 0.00$	1	1	1.183	1.480	1.545	0.990	1.387	1.469
	1	2	3.174	4.078	4.274	2.719	3.913	4.158
	1	3	6.666	8.698	9.136	5.789	8.455	8.998
	2	1	3.174	4.078	4.274	2.719	3.913	4.158
	2	2	4.733	5.918	6.179	3.959	5.547	5.877
	2	3	7.927	10.033	10.494	6.702	9.507	10.088
	3	1	6.666	8.698	9.136	5.789	8.455	8.998
	3	2	7.927	10.033	10.494	6.702	9.507	10.088
	3	3	10.650	13.317	13.903	8.908	12.481	13.224
$(\ell/a)^2 = 0.05$	1	1	1.888	2.132	2.189	1.625	1.999	2.082
	1	2	7.135	7.851	8.020	6.267	7.517	7.798
	1	3	19.151	21.790	22.401	16.758	21.088	22.038
	2	1	6.159	7.642	7.969	5.338	7.335	7.754
	2	2	12.522	13.594	13.849	11.354	12.848	13.195
	2	3	27.962	28.731	28.920	26.420	27.595	27.882
	3	1	16.487	21.238	22.268	14.343	20.638	21.931
	3	2	23.449	27.703	28.668	20.521	26.311	27.569
	3	3	39.910	43.249	44.044	36.312	40.902	41.971
$(\ell/a)^2 = 0.10$	1	1	2.333	2.614	2.679	2.023	2.452	2.548
	1	2	9.435	10.295	10.498	8.335	9.862	10.208
	1	3	26.104	29.530	30.326	22.893	28.582	29.835
	2	1	8.058	9.998	10.426	6.985	9.597	10.145
	2	2	16.795	18.229	18.570	15.235	17.230	17.694
	2	3	38.201	39.241	39.497	36.113	37.695	38.080
	3	1	22.313	28.742	30.136	19.411	27.930	29.680
	3	2	32.025	37.834	39.152	28.026	35.933	37.652
	3	3	54.997	59.596	60.691	50.041	56.362	57.835

Table 5: Dimensionless frequncies $\bar{\omega}$ of antisymmetric cross-ply laminates

E_1/E_2	25		40	
	$(-45/45)$	$(-45/45)_4$	$(45/-45)$	$(45/-45)_3$
Reddy [52]	12,357	20,154	14,636	24,825
$(\ell/a)^2 = 0.00$	12,358	20,154	14,636	24,825
$(\ell/a)^2 = 0.05$	17,419	28,409	20,631	34,994
$(\ell/a)^2 = 0.10$	21,311	34,756	25,241	42,812

Table 6: Dimensionless frequncies $\bar{\omega}$ of antisymmetric angle-ply laminates

theory, even if for the whole study they show greater magnitude, while they display an increasing trend for values around 1.2 onwards. It is also possible to say that, $\bar{\omega}$ has greater magnitude as the number of the layer of the plate increases accounting for the same thickness and for lower E_1/E_2 ratios. Finally, as previously demonstratde dimensionless fundamental frequency increases as $(\ell/a)^2$ rises.

4.3. Antisymmetric angle-ply

The last step of the present paper is focused on the analysis of the antisymmetric angle-ply laminates in terms of dimensionless frequency. Firstly, three different layouts of squared plate are considered: $(-45/45)$, $(-45/45)_4$, $(45/-45)$ and $(45/-45)_4$. The material properties for the first two columns are: $E_1/E_2 = 25$, $\nu_{12} = 0.25$, $G_{12} = G_{13} = 0.5E_2$ and for the last two $E_1/E_2 = 40$, $\nu_{12} = 0.25$, $G_{12} = G_{13} = 0.6E_2$. The frequency is dimensionless as following: $\bar{\omega} = \omega a^2/h\sqrt{\rho/E_2}$ and $n = m = 1$ is considered.

In the first two rows of Tab. 6, the comparison with Reddy [52] for the classical theory was made showing perfect agreement. The third and fourth rows display the extension to the second-order strain gradient theory, for which outcomes have a rising trend with $(\ell/a)^2$ for each case.

Finally, trends by changing a/b (increased from three to five) are drawn in Fig. 11, for $m, n = 1, 2, 3$, in terms of dimensionless fundamental frequency: $\bar{\omega} = \omega b^2/\pi^2\sqrt{\rho h/D_{22}}$. Material properties chosen as E_1/E_2 equal to 25 and 40, $\nu_{12} = 0.25$, $G_{12} = G_{13} = 0.5E_2$. The plates configurations that are studied are: $(-45/45)$, $(-45/45)_2$ and $(-45/45)_3$.

In the graph 11 trends similar to the cross-ply case are shown, on the other hand much higher magnitudes of dimensionless fundamental frequencies are reached in the present case. Decreasing in the early phase and then constant behavior is shown for the classical theory ($(\ell/a)^2 = 0.00$), on the contrary a growing behavior by changing a/b if non local effects are taken into account is observed for $(\ell/a)^2 = 0.05$ and $(\ell/a)^2 = 0.10$. It is also displayed as the magnitude of $\bar{\omega}$ grows for a major number of layers in the plate configuration to parity of thickness. The last observation regards the materials, in fact it is clear as $\bar{\omega}$ illustrates a slight more significant impact if the ratio E_1/E_2 is equal to 25.

5. Conclusion

In the present paper, the stability and dynamic analysis of simply supported nano plates are examined, applying the Kirchhoff theory and Navier solution method. An assortment of plate layouts, materials and geometries are involved, comparisons for the classical case wherever it was possible are provided, then outcomes are extended to the second-order strain gradient theory, thus taking into account nonlocal effects.

Firstly, making an analogy between laminates, for both cross- and angle-ply, in uniaxial and biaxial cases, it is clear that it is possible to exploit the higher resistance of orthotropic angle-ply plates against buckling issues compared to the cross-ply plates. Moreover, for what concerns the dimensions of the laminate in order to avoid early collapse it is needed to avoid a/b close to one, when higher order theory is employed to catch the nano plates behavior. Also, in the same material and geometrical conditions, it is preferable to use plates made by more layers to parity of plate thickness.

Finally, the procedure applied for the dynamic analysis shows non linear trends for cross- and angle-ply laminates by changing plate aspect ratios. Moreover, in this case it is also shown the much higher magnitude in terms of dimensionless fundamental frequency if angle-ply plates are employed. Also, for a defined plate thickness, the application of a greater number of layers, as well as the use of material with lower ratio E_1/E_2 induces to more considerable values of dimensioless frequencey of the structure.

In conclusion, from this study comes out that the performance of the second-order strain gradient theory is differerent from the classical one on by far, and the gap increases with plate aspect ratio and with non local ratio, thus nano plates need to be analyzed by considering non local effects.

Appendix A. Appendix

Differential operators of the Hamilton's Principle are explicitly given below, where f indicates the generic derivative operator to be applied for the partial derivation (for instance $f_{,x} = \frac{\partial}{\partial x}$):

$$\mathcal{T}_{11} = A_{11} f_{,x} + A_{16} f_{,y} - \ell^2 \left[A_{11} \left(f_{,xxx} + f_{,xyy} \right) + A_{16} \left(f_{,xyy} + f_{,yyy} \right) \right] \quad \text{(A.1)}$$

$$\mathcal{T}_{12} = A_{12} f_{,y} + A_{16} f_{,x} - \ell^2 \left[A_{12} \left(f_{,yyy} + f_{,xxy} \right) + A_{16} \left(f_{,xyy} + f_{,xxx} \right) \right] \quad \text{(A.2)}$$

$$\begin{aligned} \mathcal{T}_{13} = &-\left(B_{11} f_{,xx} + B_{12} f_{,yy} + 2B_{16} f_{,xy} \right) + \ell^2 \left[B_{11} \left(f_{,xxxx} + f_{,xxyy} \right) + \right. \\ &\left. + B_{12} \left(f_{,xxyy} + f_{,yyyy} \right) + 2B_{16} \left(f_{,xxxy} + f_{,xyyy} \right) \right] \end{aligned} \quad \text{(A.3)}$$

$$\mathcal{T}_{21} = A_{16} f_{,x} + A_{66} f_{,y} - \ell^2 \left[A_{16} \left(f_{,xxx} + f_{,xyy} \right) + A_{66} \left(f_{,xxy} + f_{,yyy} \right) \right] = \mathcal{T}_{31} \quad \text{(A.4)}$$

$$\mathcal{T}_{22} = A_{26} f_{,y} + A_{66} f_{,x} - \ell^2 \left[A_{26} \left(f_{,yyy} + f_{,xxy} \right) + A_{66} \left(f_{,xyy} + f_{,xxx} \right) \right] = \mathcal{T}_{32} \quad \text{(A.5)}$$

$$\begin{aligned} \mathcal{T}_{23} = &-\left(B_{16} f_{,xx} + B_{26} f_{,yy} + 2B_{66} f_{,xy} \right) + \ell^2 \left[B_{16} \left(f_{,xxxx} + f_{,xxyy} \right) + \right. \\ &\left. + B_{26} \left(f_{,xxyy} + f_{,yyyy} \right) + 2B_{66} \left(f_{,xxxy} + f_{,xyyy} \right) \right] = \mathcal{T}_{33} \end{aligned} \quad \text{(A.6)}$$

$$\mathcal{T}_{31} = A_{16} f_{,x} + A_{66} f_{,y} - \ell^2 \left[A_{16} \left(f_{,xxx} + f_{,xyy} \right) + A_{66} \left(f_{,xxy} + f_{,yyy} \right) \right] \quad \text{(A.7)}$$

$$\mathcal{T}_{32} = A_{26} f_{,y} + A_{66} f_{,x} - \ell^2 \left[A_{26} \left(f_{,yyy} + f_{,xxy} \right) + A_{66} \left(f_{,xyy} + f_{,xxx} \right) \right] \quad \text{(A.8)}$$

$$\mathcal{T}_{33} = -\left(B_{16}f_{,xx} + B_{26}f_{,yy} + 2B_{66}f_{,xy}\right) + \ell^2\left[B_{16}\left(f_{,xxxx} + f_{,xxyy}\right) + \right. \\ \left. + B_{26}\left(f_{,xxyy} + f_{,yyyy}\right) + 2B_{66}\left(f_{,xxxy} + f_{,xyyy}\right)\right] \tag{A.9}$$

$$\mathcal{T}_{41} = A_{12}f_{,x} + A_{26}f_{,y} - \ell^2\left[A_{12}\left(f_{,xxx} + f_{,xyy}\right) + A_{26}\left(f_{,xxy} + f_{,yyy}\right)\right] \tag{A.10}$$

$$\mathcal{T}_{42} = A_{22}f_{,y} + A_{26}f_{,x} - \ell^2\left[A_{22}\left(f_{,yyy} + f_{,xxy}\right) + A_{26}\left(f_{,xxy} + f_{,yyy}\right)\right] \tag{A.11}$$

$$\mathcal{T}_{43} = -\left(B_{12}f_{,xx} + B_{22}f_{,yy} + 2B_{26}f_{,xy}\right) + \ell^2\left[B_{12}\left(f_{,xxxx} + f_{,xxyy}\right) + \right. \\ \left. + B_{22}\left(f_{,xxyy} + f_{,yyyy}\right) + 2B_{26}\left(f_{,xyyy} + f_{,xxxy}\right)\right] \tag{A.12}$$

$$\mathcal{T}_{51} = -\left(B_{11}f_{,x} + B_{16}f_{,y}\right) + \ell^2\left[B_{11}\left(f_{,xxx} + f_{,xyy}\right) + B_{16}\left(f_{,xxy} + f_{,yyy}\right)\right] \tag{A.13}$$

$$\mathcal{T}_{52} = -\left(B_{12}f_{,y} + B_{16}f_{,x}\right) + \ell^2\left[B_{12}\left(f_{,yyy} + f_{,xxy}\right) + B_{16}\left(f_{,xyy} + f_{,xxx}\right)\right] \tag{A.14}$$

$$\mathcal{T}_{53} = D_{11}f_{,xx} + D_{12}f_{,yy} + 2D_{16}f_{,xy} - \ell^2\left[D_{11}\left(f_{,xxxx} + f_{,xxyy}\right) + \right. \\ \left. + D_{12}\left(f_{,xxyy} + f_{,yyyy}\right) + 2D_{16}\left(f_{,xyyy} + f_{,xxxy}\right)\right] \tag{A.15}$$

$$\mathcal{T}_{61} = -\left(B_{12}f_{,x} + B_{26}f_{,y}\right) + \ell^2\left[B_{12}\left(f_{,xxx} + f_{,xyy}\right) + B_{26}\left(f_{,xxy} + f_{,yyy}\right)\right] \tag{A.16}$$

$$\mathcal{T}_{62} = -\left(B_{22}f_{,y} + B_{26}f_{,x}\right) + \ell^2\left[B_{22}\left(f_{,yyy} + f_{,xxy}\right) + B_{26}\left(f_{,xyy} + f_{,xxx}\right)\right] \tag{A.17}$$

$$\begin{aligned} \mathcal{T}_{63} = D_{12}f_{,xx} + D_{22}f_{,yy} + 2D_{26}f_{,xy} - \ell^2\Big[D_{12}\left(f_{,xxxx} + f_{,xxyy}\right) \\ + D_{22}\left(f_{,xxyy} + f_{,yyyy}\right) + 2D_{26}\left(f_{,xyyy} + f_{,xxxy}\right)\Big] \end{aligned} \tag{A.18}$$

$$\begin{aligned} \mathcal{T}_{71} = 2\Big[-\left(B_{16}f_{,x} + B_{66}f_{,y}\right) + \ell^2\Big(B_{16}\left(f_{,xxx} + f_{,xyy}\right) \\ + B_{66}\left(f_{,xxy} + f_{,yyy}\right)\Big)\Big] \end{aligned} \tag{A.19}$$

$$\begin{aligned} \mathcal{T}_{72} = 2\Big[-\left(B_{26}f_{,y} + B_{66}f_{,x}\right) + \ell^2\Big(B_{26}\left(f_{,yyy} + f_{,xxy}\right) \\ + B_{66}\left(f_{,xyy} + f_{,xxx}\right)\Big)\Big] \end{aligned} \tag{A.20}$$

$$\begin{aligned} \mathcal{T}_{73} = 2\Big[D_{16}f_{,xx} + D_{26}f_{,yy} + 2D_{66}f_{,xy} - \ell^2\Big(D_{16}\left(f_{,xxxx} + f_{,xxyy}\right) \\ + D_{26}\left(f_{,xxyy} + f_{,yyyy}\right) + 2D_{66}\left(f_{,xyyy} + f_{,xxxy}\right)\Big] \end{aligned} \tag{A.21}$$

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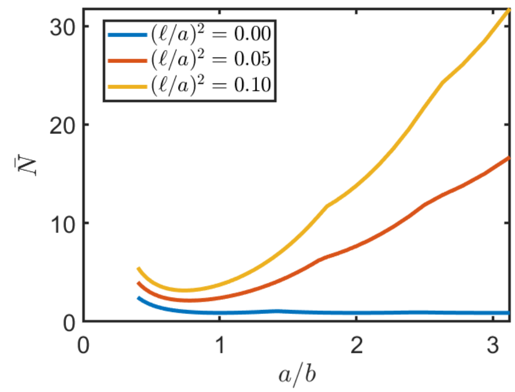
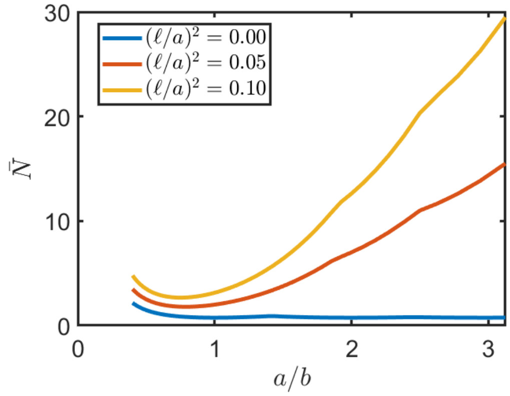
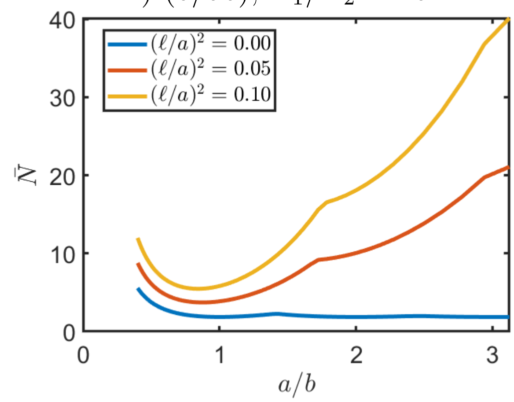


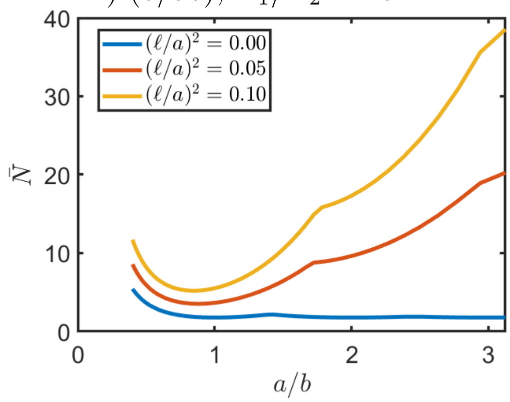
a) (0/90), $E_1/E_2 = 25$



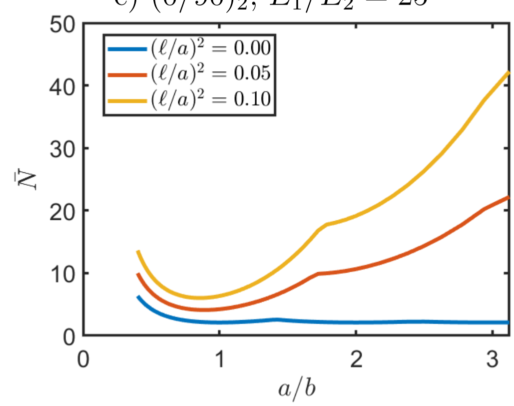
b) (0/90), $E_1/E_2 = 40$



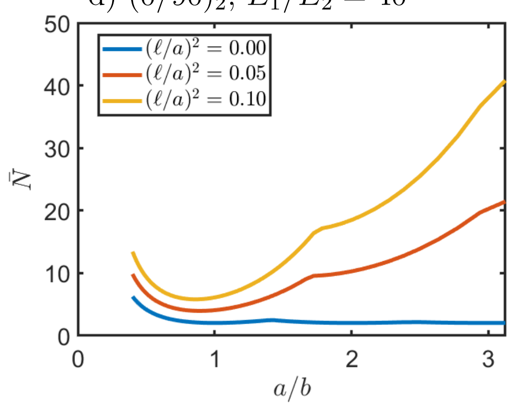
c) $(0/90)_2$, $E_1/E_2 = 25$



d) $(0/90)_2$, $E_1/E_2 = 40$



e) $(0/90)_4$, $E_1/E_2 = 25$



f) $(0/90)_4$, $E_1/E_2 = 40$

Figure 4: Uniaxial buckling load versus aspect ratio

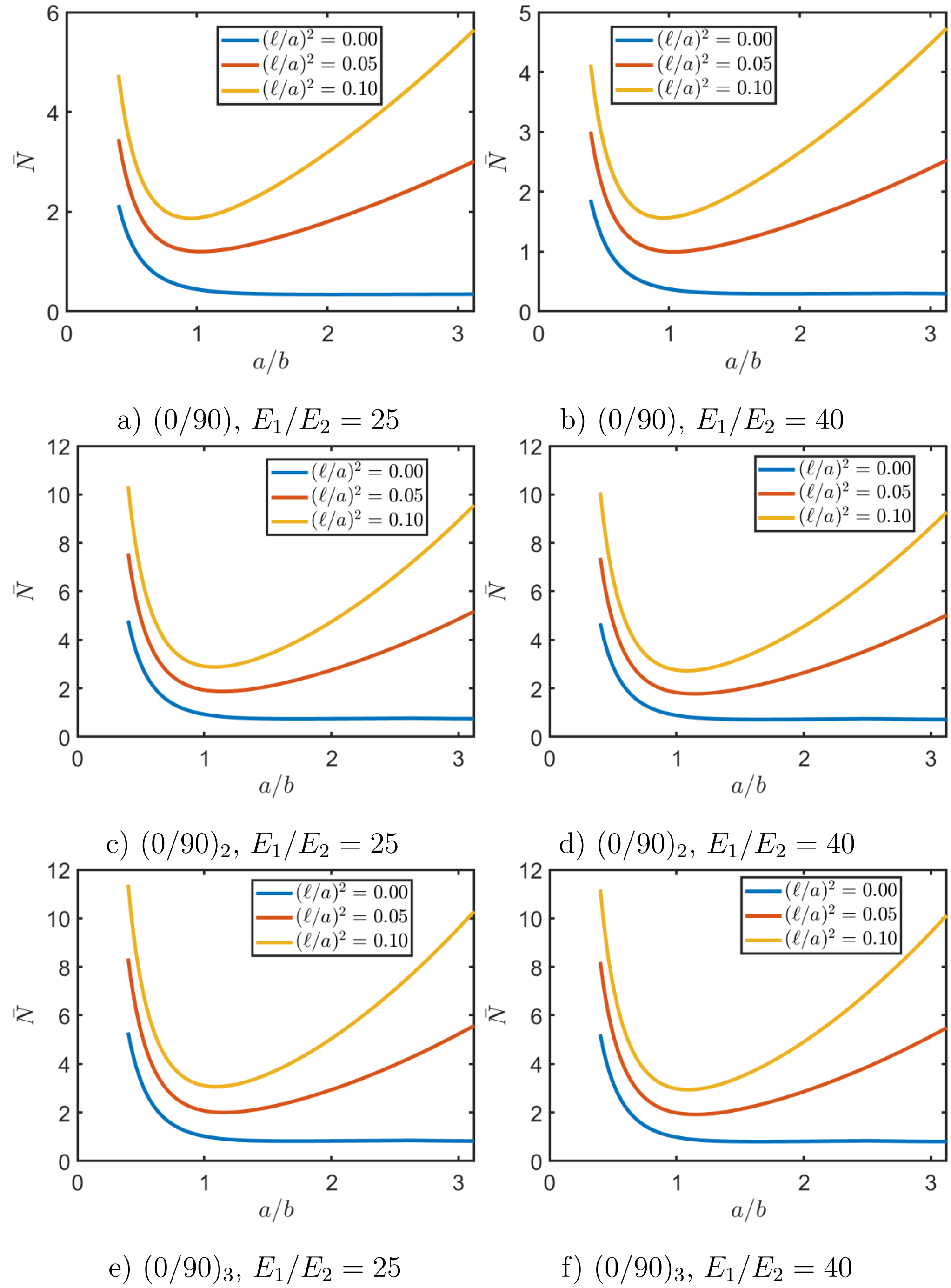
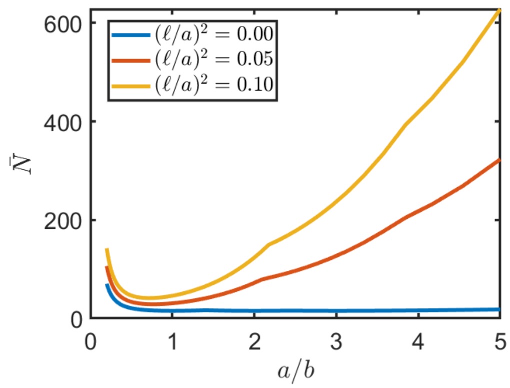
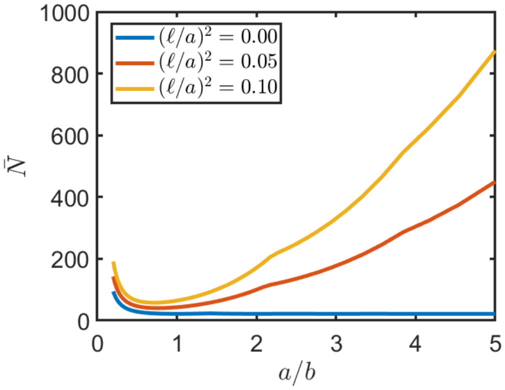


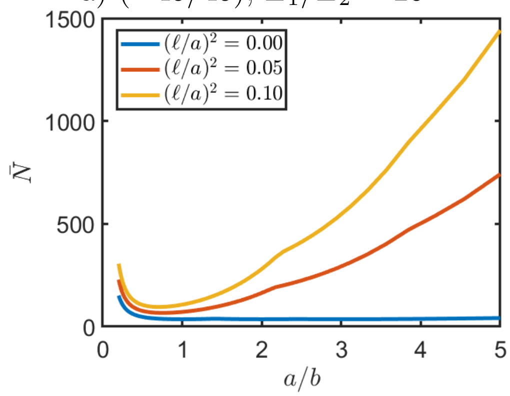
Figure 5: Biaxial buckling load versus aspect ratio



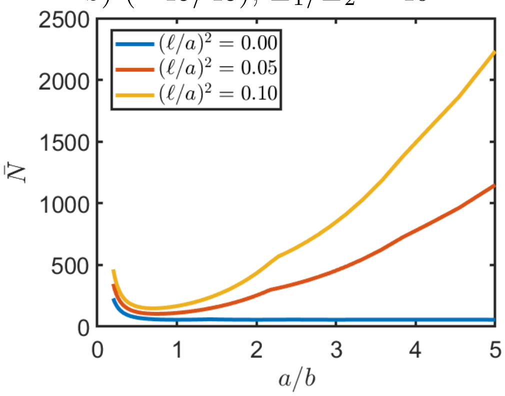
a) $(-45/45)$, $E_1/E_2 = 25$



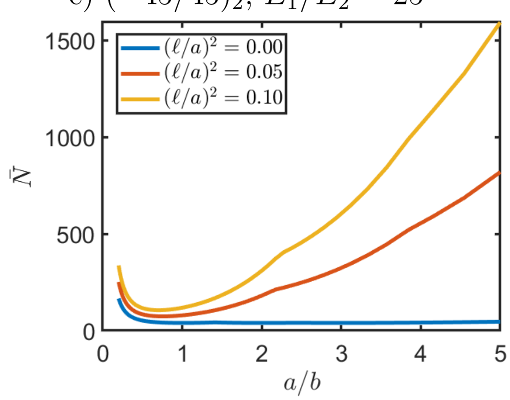
b) $(-45/45)$, $E_1/E_2 = 40$



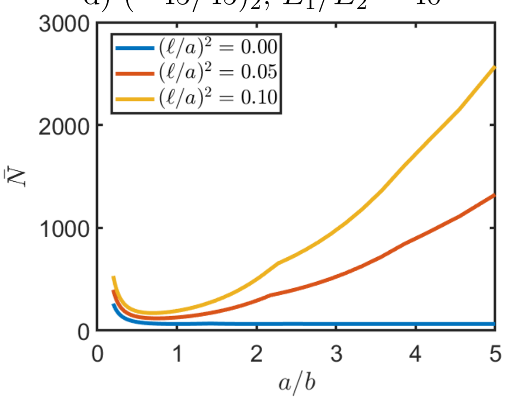
c) $(-45/45)_2$, $E_1/E_2 = 25$



d) $(-45/45)_2$, $E_1/E_2 = 40$

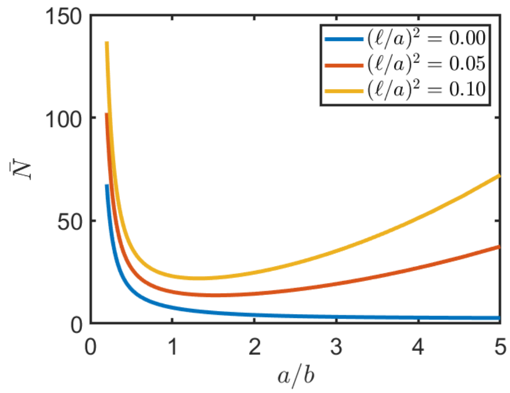


e) $(-45/45)_3$, $E_1/E_2 = 25$

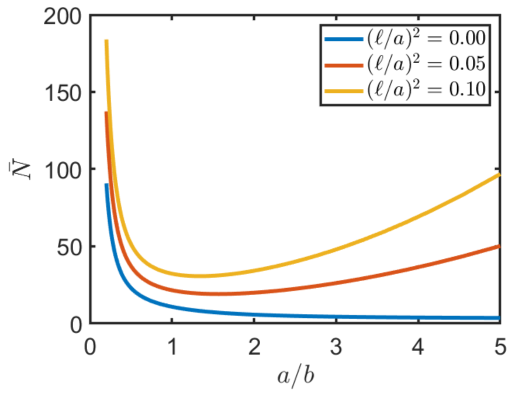


f) $(-45/45)_3$, $E_1/E_2 = 40$

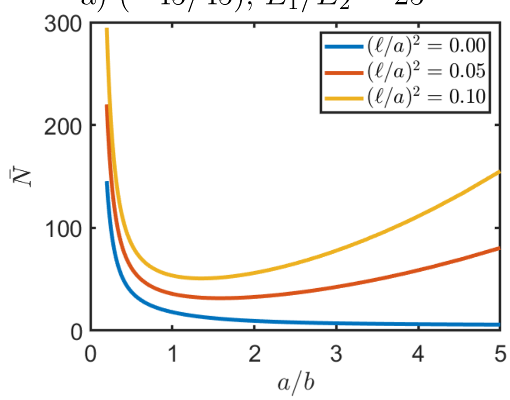
Figure 6: Uniaxial buckling load versus aspect ratio



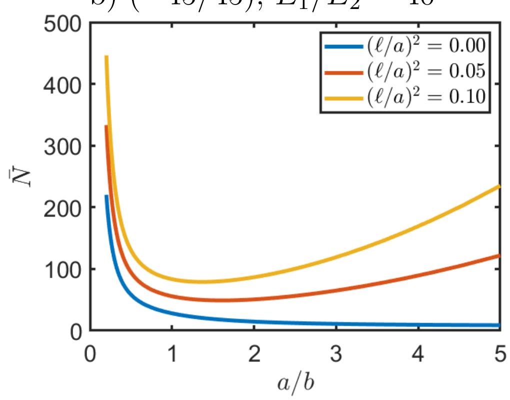
a) $(-45/45)$, $E_1/E_2 = 25$



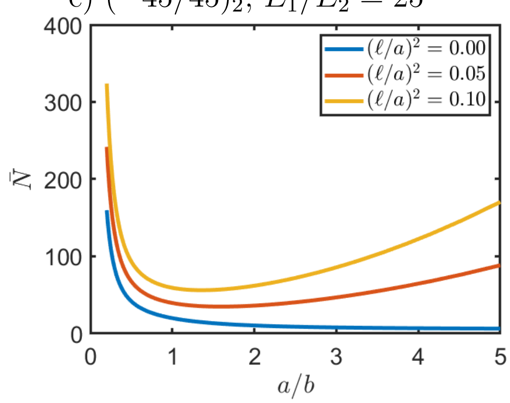
b) $(-45/45)$, $E_1/E_2 = 40$



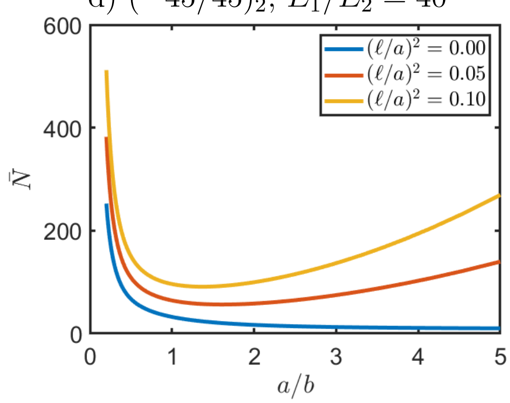
c) $(-45/45)_2$, $E_1/E_2 = 25$



d) $(-45/45)_2$, $E_1/E_2 = 40$



e) $(-45/45)_3$, $E_1/E_2 = 25$



f) $(-45/45)_3$, $E_1/E_2 = 40$

Figure 7: Biaxial buckling load versus aspect ratio

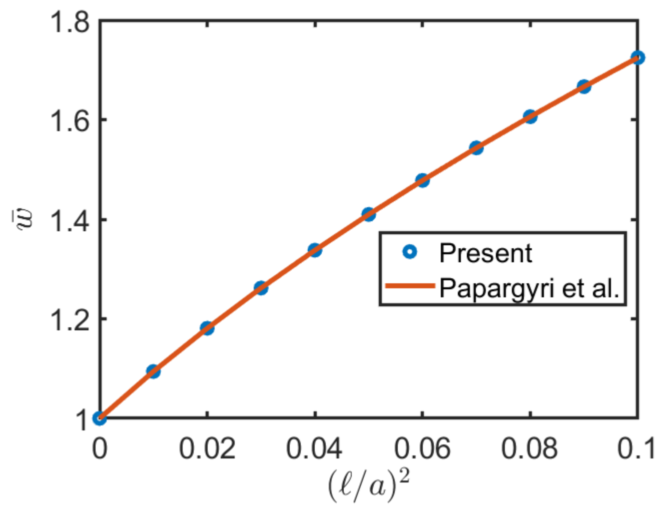


Figure 8: Vibrations - comparison with ref. [50].

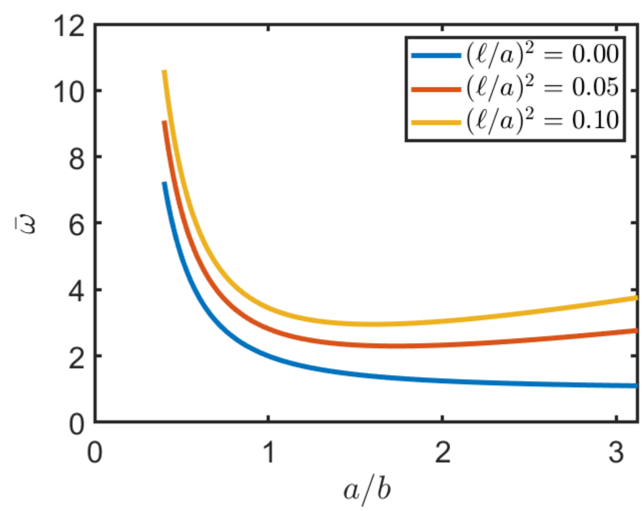
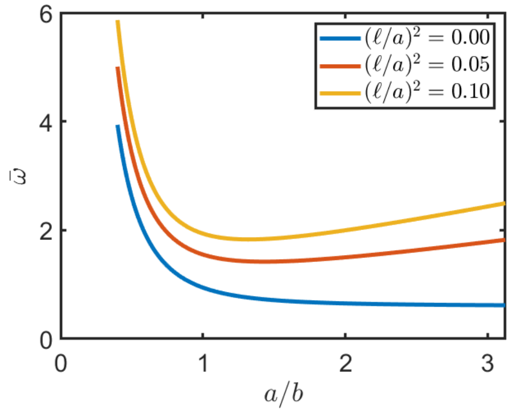
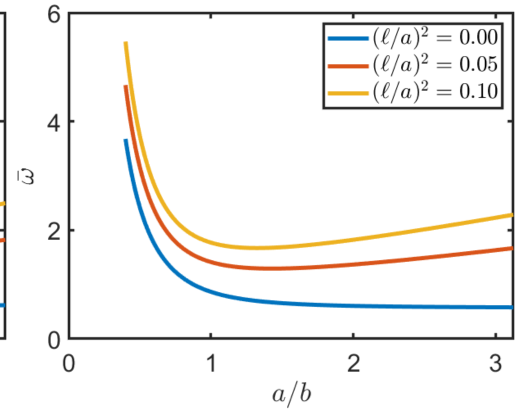


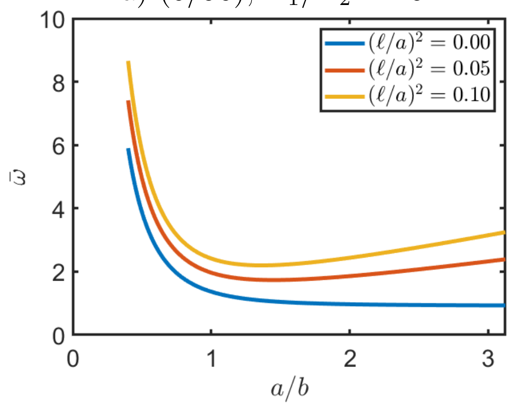
Figure 9: Nondimensionalized fundamental frequency load versus plate aspect ratio for isotropic lamina



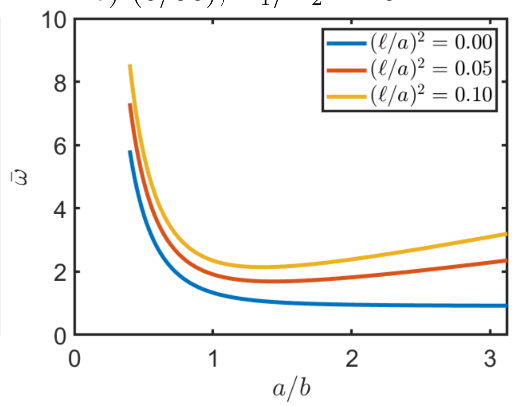
a) (0/90), $E_1/E_2 = 25$



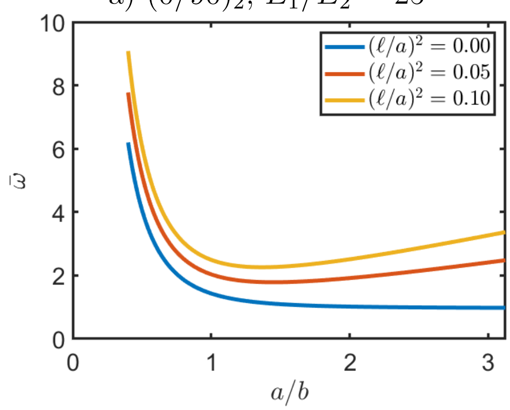
b) (0/90), $E_1/E_2 = 40$



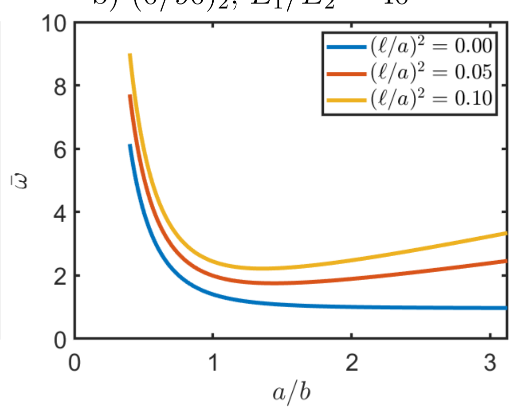
a) $(0/90)_2$, $E_1/E_2 = 25$



b) $(0/90)_2$, $E_1/E_2 = 40$

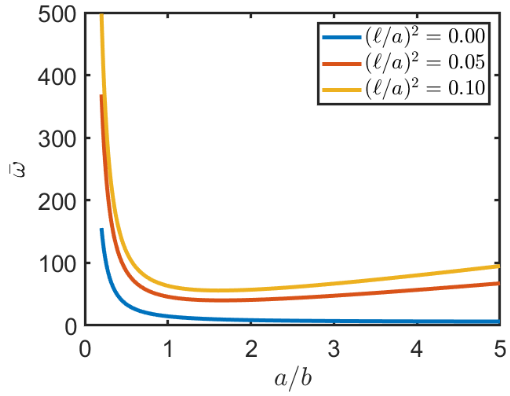


a) $(0/90)_3$, $E_1/E_2 = 25$

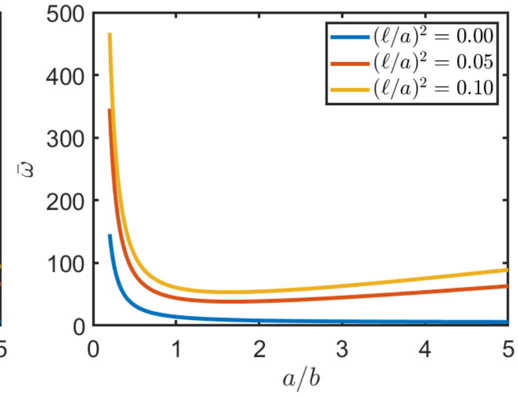


b) $(0/90)_3$, $E_1/E_2 = 40$

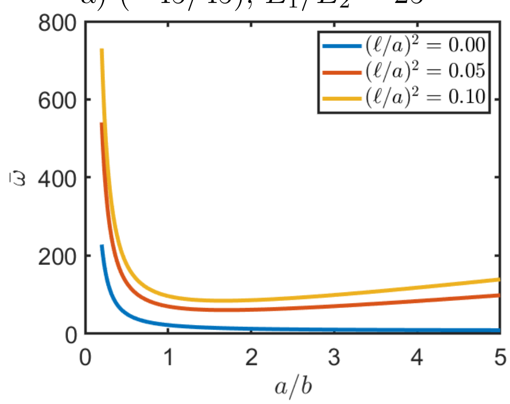
Figure 10: Dimensionless fundamental frequency versus plate aspect ratio for antisymmetric cross-ply laminates to changing non local ratios



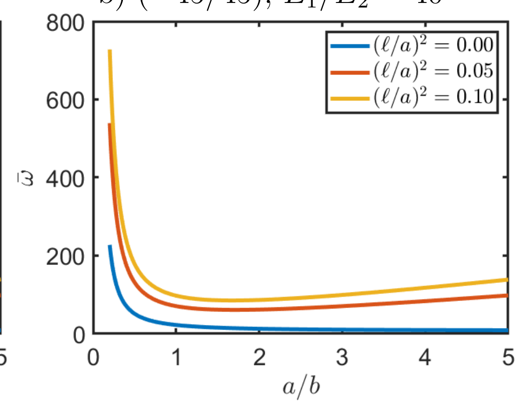
a) $(-45/45)$, $E_1/E_2 = 25$



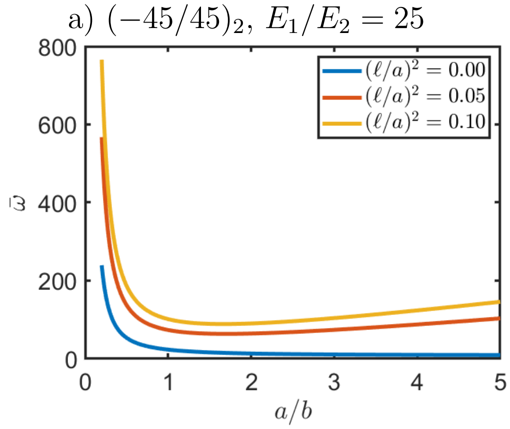
b) $(-45/45)$, $E_1/E_2 = 40$



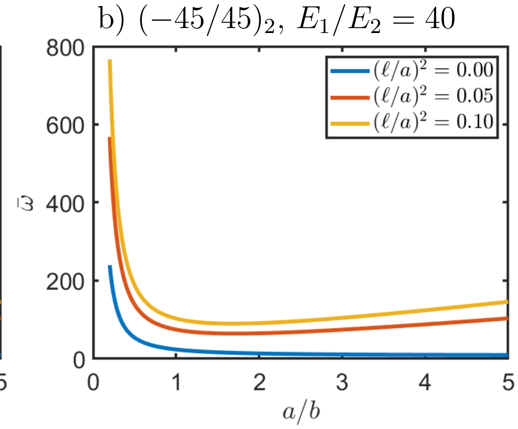
a) $(-45/45)_2$, $E_1/E_2 = 25$



b) $(-45/45)_2$, $E_1/E_2 = 40$



a) $(-45/45)_3$, $E_1/E_2 = 25$



b) $(-45/45)_3$, $E_1/E_2 = 40$

Figure 11: Dimensionless fundamental frequency versus plate aspect ratio for antisymmetric angle-ply laminates to changing non local ratios